*MAT_162
*MAT_COMPOSITE_MSC_DMG

A PROGRESSIVE COMPOSITE DAMAGE MODEL FOR UNIDIRECTIONAL AND WOVEN FABRIC COMPOSITES

UD-CCM Updates on MAT162 USER MANUAL
Version 22A-2022

Materials Sciences Corporation (MSC)
&
University of Delaware Center for Composite Materials (UD-CCM)

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*MAT_COMPOSITE_MSC_{*OPTION*}

Available options include:

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DMG

These are Material Types 161 and 162. These material types may be used to model the progressive failure in composite materials consisting of unidirectional and woven fabric layers subjected to high strain-rate and high pressure loading conditions. The progressive layers failure criteria have been established by adopting the methodology developed by Hashin [1980] with a generalization to include the effect of highly constrained pressure on composite failure. These failure models can be used to effectively simulate fiber failure, matrix damage, and delamination behavior under all conditions – opening, closure, and sliding of failure surfaces. The model with DMG option (material 162) is a generalization of the basic layer failure model of Material 161 by adopting the damage mechanics approach [Matzenmiller et al., 1995] for characterizing the softening behavior after damage initiation. These models require an additional license from Materials Sciences Corporation, which developed and supports these models in collaboration with University of Delaware Center for Composite Materials (UD-CCM).

Card 1	1	2	3	4	5	6	7	8
Variable	MID	RO	EA	EB	EC	PRBA	PRCA	PRCB
Type	A8	F	F	F	F	F	F	F

Card 2	1	2	3	4	5	6	7	8
Variable	GAB	GBC	GCA	AOPT	MACF			
Type	F	F	F	F	I			

Card 3	1	2	3	4	5	6	7	8
Variable	XP	YP	ZP	A1	A2	A3		
Type	F	F	F	F	F	F		

Card 4	1	2	3	4	5	6	7	8
Variable	V1	V2	V3	D1	D2	D3	BETA	
Type	F	F	F	F	F	F	F	

Card 5	1	2	3	4	5	6	7	8
Variable	SAT	SAC	SBT	SBC	SCT	SFC	SFS	S_AB
Type	F	F	F	F	F	F	F	F

Card 6	1	2	3	4	5	6	7	8
Variable	S_BC	S_CA	SFFC	AMODEL	PHIC	E_LIMT	S_DELM	
Type	F	F	F	F	F	F	F	

Card 7	1	2	3	4	5	6	7	8
Variable	OMGMX	ECRSH	EEXPN	CERATE1	AM1			
Type	F	F	F	F	F			

Define the following card if and only if the option DMG is specified.

Card 8	1	2	3	4	5	6	7	8
Variable	AM2	AM3	AM4	CERATE2	CERATE3	CERATE4		
Type	F	F	F	F	F	F		

VARIABLE	DESCRIPTION
MID	Material identification. A unique number or label not exceeding 8 characters must be specified.
RO	Mass density
EA	E_a, Young’s modulus - longitudinal direction♥
EB	E_b, Young’s modulus - transverse direction♥
EC	E_c, Young’s modulus – through thickness direction♥
PRBA	ν_{ba}, Poisson’s ratio ba
PRCA	ν_{ca}, Poisson’s ratio ca
PRCB	ν_{cb}, Poisson’s ratio cb
GAB	G_{ab}, shear modulus ab
GBC	G_{bc}, shear modulus bc
GCA	G_{ca}, shear modulus ca

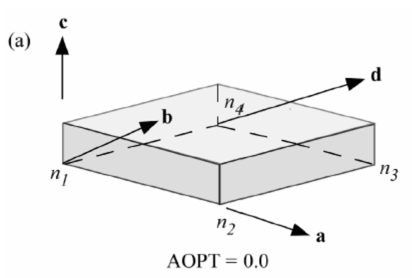
♥ LS-DYNA® notations a, b, & c have the same meaning as for orthotropic material axes notations 1, 2, & 3.

VARIABLE	DESCRIPTION
AOPT	Material axes option, see Figure 2.1. (in KEYWORD Manual) **EQ.0.0**: locally orthotropic with material axes determined by element nodes as shown in Figure 2.1. Nodes 1, 2, and 4 of an element are identical to the Nodes used for the definition of a coordinate system by *DEFINE_COORDINATE_NODES. **EQ.1.0**: locally orthotropic with material axes determined by a point in space and the global location of the element center, this is the a-direction. **EQ.2.0**: globally orthotropic with material axes determined by vectors defined below, as with *DEFINE_COORDINATE_VECTOR. **LT.0.0**: the absolute value of AOPT is a coordinate system ID number (CID on *DEFINE_COORDINATE_NODES, *DEFINE_COORDINATE_SYSTEM or *DEFINE_COORDINATE_VECTOR). Available in R3 version of 971 and later.
MACF	Material axes change flag: EQ.1: No change, default, EQ.2: switch material axes a & b, EQ.3: switch material axes a & c, EQ.4: switch material axes b & c.
XP YP ZP	Define coordinates of point **p** for AOPT = 1.
A1 A2 A3	Define components of vector **a** for AOPT = 2.
V1 V2 V3	Define components of vector **v** for AOPT = 3.
D1 D2 D3	Define components of vector **d** for AOPT = 2.
BETA	Layer in-plane rotational angle in degrees.
SAT	Longitudinal tensile strength, S_{aT}
SAC	Longitudinal compressive strength, S_{aC}
SBT	Transverse tensile strength, S_{bT}
SBC	Transverse compressive strength, S_{bC}
SCT	Through thickness tensile strength, S_{cT}
SFC	Crush strength, S_{FC}
SFS	Fiber mode shear strength, S_{FS}

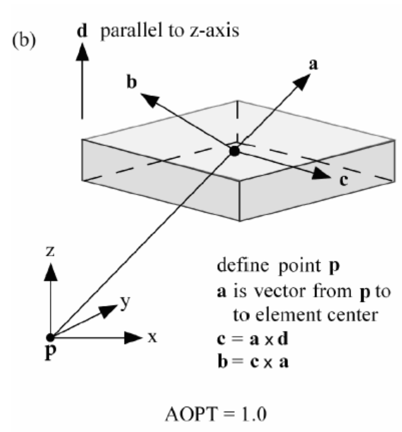
VARIABLE	DESCRIPTION
S_AB♦	Matrix mode shear strength, ab plane, see below, S_{ab}
S_BC♦	Matrix mode shear strength, bc plane, see below, S_{bc}
S_CA♦	Matrix mode shear strength, ca plane, see below, S_{ca}
SFFC	Scale factor for residual compressive strength, S_{FFC}
AMODEL	Material models: EQ.1: Unidirectional lamina model EQ.2: Fabric lamina model
PHIC	Coulomb friction angle for matrix and delamination failure, $\varphi < 90^{\circ}$
S_DELM	Scale factor for delamination criterion, S
OMGMX	Limit damage parameter for elastic modulus reduction, $\varpi_{\max}$
E_LIMT	Element eroding axial strain
ECRSH	Limit compressive relative volume for element eroding
EEXPN	Limit expansive relative volume for element eroding
CERATE1	Coefficient for strain rate dependent strength properties, C_{rate1}
CERATE2	Coefficient for strain rate dependent axial moduli, C_{rate2}
CERATE3	Coefficient for strain rate dependent shear moduli, C_{rate3}
CERATE4	Coefficient for strain rate dependent transverse moduli, C_{rate4}
AM1	Coefficient for strain softening property for fiber damage in a direction, m_1
AM2	Coefficient for strain softening property for transverse compressive matrix failure mode in b direction (unidirectional) or for fiber damage mode in b direction (fabric), m_2
AM3	Coefficient for strain softening property for fiber crush and punch shear damage, m_3
AM4	Coefficient for strain softening property for matrix failure and delamination damage, m_4

♦ LS-DYNA KEYWORD manual presents these parameters as: SAB, SBC, & SCA. Since SBC is also used to define the “transverse compressive strength,” we have used S_xx to represent shear strength in xx plane.

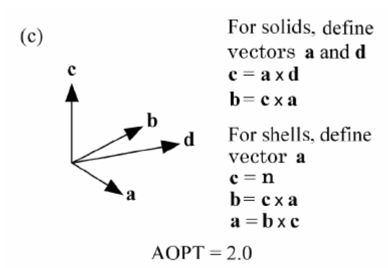
Figure 2.1. from the KEYWORD Manual of LS-DYNA® is reproduced here for the convenience of the MAT162 users.



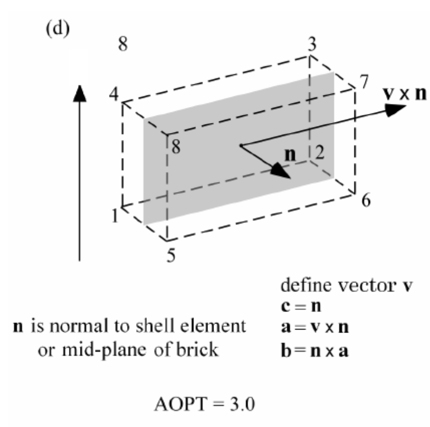
(a) AOPT = 0.0



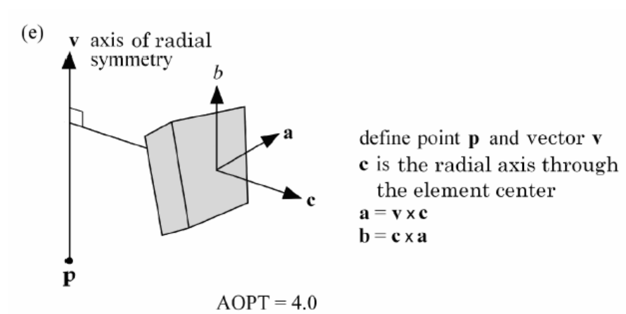
(b) AOPT = 1.0



(c) AOPT = 2.0



(d) AOPT = 3.0



(e) AOPT = 4.0

Figure 2.1. Options for determining principal material axes: (a) AOPT = 0.0, (b) AOPT = 1.0 for brick elements, (c) AOPT = 2.0, (d) AOPT = 3.0, and (e) AOPT=4.0 for brick elements.

Figure 1: Material Axes Definition presented in the LS-DYNA KEYWORD Manual (OLDER VERSIONS)

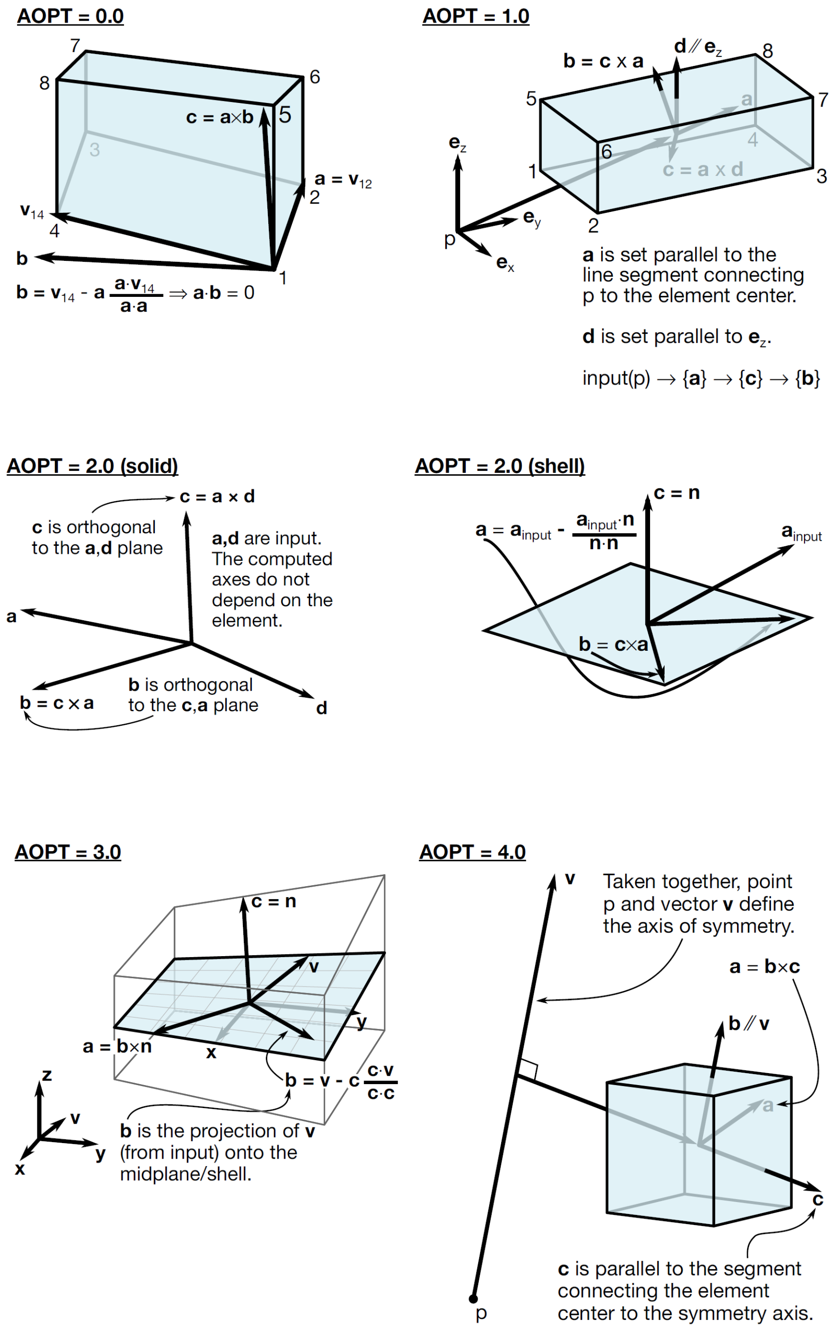


Figure 1 (Continued): Material Axes Definition presented in the LS-DYNA Manuals (NEW VERSIONS, May 2014)

MATERIAL MODELS

Failure models based on the 3D stresses/strains in a composite lamina with improved progressive failure modeling capability are established for a unidirectional and for a fabric composite lamina. ***While the LS-DYNA KEYWORD manual presents the stress based formulations, this manual presents the strain based formulations.*** These models can be used to effectively simulate the fiber failure, matrix failure, and delamination behavior of composites under high strain-rate and high pressure ballistic impact conditions.

The unidirectional and fabric lamina failure criteria and the associated property degradation models are described as follows. All the failure criteria are expressed in terms of stress components based on ply level strains $(\varepsilon_1, \varepsilon_2, \varepsilon_3, \varepsilon_{12}, \varepsilon_{23}, \varepsilon_{31}) = (\varepsilon_a, \varepsilon_b, \varepsilon_c, \varepsilon_{ab}, \varepsilon_{bc}, \varepsilon_{ca})$. The associated elastic moduli are $(E_1, E_2, E_3, G_{12}, G_{23}, G_{31}) = (E_a, E_b, E_c, G_{ab}, G_{bc}, G_{ca})$. Note that for the unidirectional model, a, b, and c denote the fiber, in-plane transverse and out-of-plane or through-thickness directions, respectively; while for the fabric model, a, b, and c denote the in-plane fill, in-plane warp and out-of-plane or through-thickness directions, respectively.

UNIDIRECTIONAL LAMINA DAMAGE MODEL

Fiber Mode Failures

The fiber failure criteria of Hashin [1980] for a unidirectional layer are generalized to characterize the fiber damage in terms of strain components for a unidirectional layer. Three damage functions are used for fiber failure, one in tension/shear, one in compression, and another one in crush under pressure. They are chosen in terms of quadratic strain forms as follows:

UNIDIRECTIONAL TENSION-SHEAR FIBER MODE 1uf along direction 1, A, a:

$$f_1 - r_1^2 = \left(\frac{E_a \langle \varepsilon_a \rangle}{S_{aT}}\right)^2 + \left(\frac{G_{ab}^2 \varepsilon_{ab}^2 + G_{ca}^2 \varepsilon_{ca}^2}{S_{FS}^2}\right) - r_1^2 = 0 \tag{1}$$

UNIDIRECTIONAL COMPRESSION FIBER MODE 2uf along direction 1, A, a:

$$f_2 - r_2^2 = \left(\frac{E_a \langle \varepsilon_a^{/} \rangle}{S_{aC}}\right)^2 - r_2^2 = 0\,, \;\rightarrow\; \varepsilon_a^{/} = -\varepsilon_a - \frac{\langle -E_c \varepsilon_c - E_b \varepsilon_b \rangle}{2E_a} \tag{2}$$

UNIDIRECTIONAL CRUSH FIBER MODE 3uf along direction 3, C, c:

$$f_3 - r_3^2 = \left(\frac{E_c \langle -\varepsilon_c \rangle}{S_{FC}} \right)^2 - r_3^{\,2} = 0 \tag{3}$$

where $\langle\ \rangle$ are Macaulay brackets, S_{aT} and S_{aC} are the tensile and compressive strengths in the fiber direction, and S_{FS} and S_{FC} are the layer strengths associated with the fiber shear and crush failure, respectively. The damage thresholds, r_j, j = 1, 2, 3, have the initial values equal to 1 before the damage initiated, and are updated due to damage accumulation in the associated damage modes.

Matrix Mode Failures

Matrix mode failures must occur without fiber failure, and hence they will be on planes parallel to fibers. Two matrix damage functions are chosen for the failure plane perpendicular and parallel to the layering planes. They have the forms:

UNIDIRECTIONAL TRANSVERSE COMPRESSION MATRIX MODE 4um along direction 2, B, b:

$$f_4 - r_4^2 = \left(\frac{E_b \langle -\varepsilon_b \rangle}{S_{bC}} \right)^2 - r_4^{\,2} = 0 \tag{4}$$

UNIDIRECTIONAL PERPENDICULAR MATRIX SHEAR MODE 5um along direction 2, B, b (tension only) and on planes 12, AB, ab & 23, BC, bc:

$$f_5 - r_5^2 = \left(\frac{E_b \langle \varepsilon_b \rangle}{S_{bT}} \right)^2 + \left(\frac{G_{bc}\varepsilon_{bc}}{S_{bc0} + S_{SRB}} \right)^2 + \left(\frac{G_{ab}\varepsilon_{ab}}{S_{ab0} + S_{SRB}} \right)^2 - r_5^{\,2} = 0 \tag{5}$$

UNIDIRECTIONAL PARALLEL MATRIX MODE (DELAMINATION) 6um along direction 3, C, c (tension only) and on planes 23, BC, bc & 31, CA, ca:

$$f_6 - r_6^2 = S^2 \left\{ \left(\frac{E_c \langle \varepsilon_c \rangle}{S_{cT}} \right)^2 + \left(\frac{G_{bc}\varepsilon_{bc}}{S_{bc0} + S_{SRC}} \right)^2 + \left(\frac{G_{ca}\varepsilon_{ca}}{S_{ca0} + S_{SRC}} \right)^2 \right\} - r_6^{\,2} = 0 \tag{6}$$

where S_{bT} and S_{cT} are the transverse tensile strengths of the corresponding tensile modes ($\varepsilon_b > 0$ or $\varepsilon_c > 0$); and S_{ab0}, S_{bc0}, & S_{ca0} are the quasi-static shear strength values. Under compressive transverse strain, $\varepsilon_b < 0$ or $\varepsilon_c < 0$, the damaged surface is considered to be “closed”, and the shear strengths are assumed to depend on the compressive normal strains based on the Mohr-Coulomb theory, i.e.:

$$S_{SRB} = E_b \tan(\varphi)\langle -\varepsilon_b \rangle$$
$$S_{SRC} = E_c \tan(\varphi)\langle -\varepsilon_c \rangle \tag{7}$$

where φ is a material constant as tan(φ) is similar to the coefficient of friction. The damage thresholds r_j, j = 4, 5, 6, have the initial values equal to 1 before the damage initiated, and are updated due to damage accumulation of the associated damage modes.

Failure predicted by the criterion of f_4 and f_5 can be referred to as transverse matrix failure, while the matrix failure predicted by f_6, which is parallel to the layer, can be referred as the delamination mode when it occurs within the elements that are adjacent to the ply interface. Note that a scale factor S is introduced to provide better correlation of delamination area with experiments. The scale factor S can be determined by fitting the analytical prediction to experimental data for the delamination area.

FABRIC LAMINA DAMAGE MODEL

Fiber Mode Failures

The fiber failure criteria of Hashin [1980] for a unidirectional layer are generalized to characterize the fiber damage in terms of strain components for a plain weave layer. The fill and warp fiber tensile/shear damage are given by the quadratic interaction between the associated axial and through the thickness shear strains, i.e.:

TENSION-SHEAR FIBER MODE 7f along direction 1, A, a (tension only), and on plane 31, CA, ca [Eq. 8a]:
TENSION-SHEAR FIBER MODE 8f along direction 2, B, b (tension only), and on plane 23, BC, bc [Eq. 8b]:

$$f_7 - r_7^2 = \left(\frac{E_a\langle\varepsilon_a\rangle}{S_{aT}}\right)^2 + \left(\frac{G_{ca}\varepsilon_{ca}}{S_{aFS}}\right)^2 - r_7^2 = 0 \tag{8a}$$

$$f_8 - r_8^2 = \left(\frac{E_b\langle\varepsilon_b\rangle}{S_{bT}}\right)^2 + \left(\frac{G_{bc}\varepsilon_{bc}}{S_{bFS}}\right)^2 - r_8^2 = 0 \tag{8b}$$

where S_{aT} and S_{bT} are the axial tensile strengths in the fill and warp directions, respectively, and S_{aFS} and S_{bFS} are the lamina shear strengths due to fiber shear failure in the fill and warp directions. These failure criteria are applicable when the associated ε_a or ε_b is positive. The damage thresholds r_7 and r_8 are equal to 1 without damage. It is assumed $S_{aFS} = S_{FS}$, and $S_{bFS} = S_{FS} \times S_{bT} / S_{aT}$.

COMPRESSION FIBER MODE 9f along direction 1, A, a [Eq. 9a]:
COMPRESSION FIBER MODE 10f along direction 2, B, b [Eq. 9b]:

When ε_a or ε_b is compressive, it is assumed that the in-plane compressive damage in the fill and warp directions are given by the maximum strain criterion, i.e.:

$$f_9 - r_9^2 = \left(\frac{E_a \left\langle \varepsilon_a^{/} \right\rangle}{S_{aC}} \right)^2 - r_9^{\,2} = 0 \;\rightarrow\; \varepsilon_a^{/} = -\varepsilon_a - \left\langle -\varepsilon_c \right\rangle \frac{E_c}{E_a} \tag{9a}$$

$$f_{10} - r_{10}^2 = \left(\frac{E_b \left\langle \varepsilon_b^{/} \right\rangle}{S_{bC}} \right)^2 - r_{10}^{\,2} = 0 \;\rightarrow\; \varepsilon_b^{/} = -\varepsilon_b - \left\langle -\varepsilon_c \right\rangle \frac{E_c}{E_b} \tag{9b}$$

where S_{aC} and S_{bC} are the axial compressive strengths in the fill and warp directions, respectively, and r_9 and r_{10} are the corresponding damage thresholds. Note that the effect of through the thickness compressive strain on the in-plane compressive damage is taken into account in the above two equations.

CRUSH FIBER MODE 11f along direction 3, C, c [Eq. 10]:

When a composite material is subjected to transverse impact by a projectile, high compressive stresses will generally occur in the impact area with high shear stresses in the surrounding area between the projectile and the target material. While the fiber shear punch damage due to the high shear stresses can be accounted for by equation (1), the crush damage due to the high through the thickness compressive pressure is modeled using the following criterion:

$$f_{11} - r_{11}^2 = \left(\frac{E_c \left\langle -\varepsilon_c \right\rangle}{S_{FC}} \right)^2 - r_{11}^{\,2} = 0 \tag{10}$$

where S_{FC} is the fiber crush strengths and r_{11} is the associated damage threshold.

Matrix Mode Failures

IN-PLANE MATRIX SHEAR MODE 12m on plane 12, AB, ab [Eq. 11]:

A plain weave layer can be damaged under in-plane shear stressing without occurrence of fiber breakage. This in-plane matrix damage mode is given by:

$$f_{12} - r_{12}^2 = \left(\frac{G_{ab}\varepsilon_{ab}}{S_{ab}}\right)^2 - r_{12}{}^2 = 0 \tag{11}$$

where S_{ab} is the layer shear strength due to matrix shear failure and r_{12} is the damage threshold.

PARALLEL MATRIX SHEAR & DELAMINATION MODE 13m along direction 3, C, c (tension only) and on planes 23, BC, bc & 31, CA, ca [Eq. 12]:

Another failure mode, which is due to the quadratic interaction between the transverse strains, is expected to be mainly a matrix failure. This through the thickness matrix failure criterion is assumed to have the following form:

$$f_{13} - r_{13}^2 = S^2\left\{\left(\frac{E_c\langle\varepsilon_c\rangle}{S_{cT}}\right)^2 + \left(\frac{G_{bc}\varepsilon_{bc}}{S_{bc0}+S_{SRC}}\right)^2 + \left(\frac{G_{ca}\varepsilon_{ca}}{S_{ca0}+S_{SRC}}\right)^2\right\} - r_{13}{}^2 = 0 \tag{12}$$

where r_{13} is the damage threshold, S_{cT} is the through the thickness tensile strength for tensile ε_c, and S_{bc0} and S_{ca0} are the quasi-static shear strengths. The damage surface due to equation (12) is parallel to the composite layering plane. Under compressive through the thickness strain, $\varepsilon_c < 0$, the damaged surface (delamination) is considered to be “closed”, and the shear strengths are assumed to depend on the compressive normal strain ε_c similar to the Mohr-Coulomb theory, i.e.:

$$S_{SRC} = E_c \tan(\varphi)\langle -\varepsilon_c\rangle \tag{13}$$

where φ is the Coulomb’s friction angle. When damage predicted by this criterion occurs within elements that are adjacent to the ply interface, the failure plane is expected to be parallel to the layering planes, and, thus, can be referred to as the delamination mode. Note that a scale factor S is introduced to provide better correlation of delamination area with experiments. The scale factor S can be determined by fitting the analytical prediction to experimental data for the delamination area.

DAMAGE PROGRESSION MODEL

A set of damage variables ϖ_i with i = 1, … , 6; are introduced to relate the onset and growth of damage to stiffness losses in the material. The compliance matrix $[S]$ is related to the damage variables as (Matzenmiller, et al., 1995):

$$[S]=\begin{bmatrix} \frac{1}{(1-\varpi_1)E_a} & \frac{-\nu_{ba}}{E_b} & \frac{-\nu_{ca}}{E_c} & 0 & 0 & 0 \\ \frac{-\nu_{ab}}{E_a} & \frac{1}{(1-\varpi_2)E_b} & \frac{-\nu_{cb}}{E_c} & 0 & 0 & 0 \\ \frac{-\nu_{ac}}{E_a} & \frac{-\nu_{bc}}{E_b} & \frac{1}{(1-\varpi_3)E_c} & 0 & 0 & 0 \\ 0 & 0 & 0 & \frac{1}{(1-\varpi_4)G_{ab}} & 0 & 0 \\ 0 & 0 & 0 & 0 & \frac{1}{(1-\varpi_5)G_{bc}} & 0 \\ 0 & 0 & 0 & 0 & 0 & \frac{1}{(1-\varpi_6)G_{ca}} \end{bmatrix} \qquad (14)$$

The stiffness matrix $[C]$ is obtained by inverting the compliance matrix, i.e., $[C]=[S]^{-1}$. As suggested in Matzenmiller, et al., (1995), the growth rate of damage variables, ϖ_i, is governed by the damage rule of the form:

$$\dot{\varpi}_i = \max\left\{\dot{\phi}_j q_{ij}\right\} \qquad (15)$$

where the scalar damage functions $\dot{\phi}_j$ control the amount of growth and the vector-valued matrix q_{ij} (i = 1,…6, j = 1, …, 13) provide the coupling between the individual damage variables (i) and the various damage modes (j). Note that there are six damage modes (j = 1, …, 6) for the "unidirectional lamina model" and seven damage modes (j = 7, …, 13) for the "fabric lamina model." The damage criteria $f_j - r_j^2 = 0$ of Eqs. (1) to (12) provide the damage surfaces in strain space for the unidirectional and fabric lamina models, respectively. Damage growth, $\dot{\phi}_j > 0$, will occur when the strain path crosses the updated damage surface $f_j - r_j^2 = 0$ and the strain increment has a non-zero component in the direction of the normal to the damage surface, i.e., $\sum_i \frac{\partial f_j}{\partial \varepsilon_i}\dot{\varepsilon}_i > 0$. Combined with damage growth functions $\gamma_j(\varepsilon_i, \varpi_i)$; $\dot{\phi}_j$ is assumed to have the form:

$$\dot{\phi}_j = \sum_i \gamma_j \frac{\partial f_j}{\partial \varepsilon_i}\dot{\varepsilon}_i \quad \text{(no summation over j)} \qquad (16)$$

Choosing

$$\gamma_j = \frac{1}{2}\left(1-\phi_j\right) f_j^{\frac{m_j}{2}-1} \tag{17}$$

and noting that

$$\sum_i \frac{\partial f_j}{\partial \varepsilon_i}\dot{\varepsilon}_i = \dot{f}_j \tag{18}$$

for the quadratic functions given by Eqs. (1) to (6) and Eqs. (8) to (12), lead to:

$$\dot{\phi}_j = \frac{1}{2}\left(1-\phi_j\right) f_j^{\frac{m_j}{2}-1} \dot{f}_j \tag{19}$$

where ϕ_j is the scalar damage function associated with the j[th] failure mode, and m_j is a material constant for softening behavior. The scalar damage function ϕ_j can be obtained by integrating Eq. (19) as follows:

$$\begin{aligned} & f_j = r_j^2 \;\rightarrow \quad \dot{f}_j = 2r_j\dot{r}_j \;\; \& \;\; f_j^{\frac{m_j}{2}-1} = r_j^{m_j-2} \\ & \dot{\phi}_j = \left(1-\phi_j\right) r_j^{m_j-1}\dot{r}_j \;\rightarrow \quad \int_0^{\phi_j} \frac{d\phi_j}{\left(1-\phi_j\right)} = \int_0^{r_j} r_j^{m_j-1} dr_j \;..\rightarrow \quad \phi_j = 1-\exp\left(\frac{1}{m_j}\left(1-r_j^{m_j}\right)\right) \end{aligned} \tag{20}$$

The damage coupling matrix q_{ij} is considered for the unidirectional and fabric lamina models as follows.

DAMAGE COUPLING MATRIX FOR UNIDIRECTIONAL LAMINA MODEL

Eq. (21) is the damage coupling matrix, and Fig. 2 illustrates how Eq. (21) is associated with the modulus reduction for the unidirectional lamina model.

$$q_{ij}^{U} = \begin{bmatrix} 1 & 1 & 1 & 0 & 0 & 0 \\ 0 & 0 & 1 & 1 & 1 & 0 \\ 0 & 0 & 1 & 0 & 0 & 1 \\ 1 & 1 & 1 & 1 & 1 & 0 \\ 0 & 0 & 1 & 1 & 1 & 1 \\ 1 & 1 & 1 & 0 & 0 & 1 \end{bmatrix} \quad i = 1, \ldots, 6;\ j = 1, \ldots, 6. \tag{21}$$

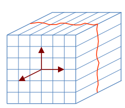
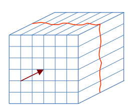
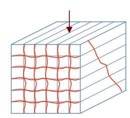
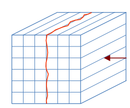
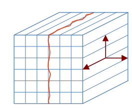
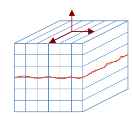
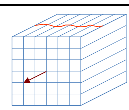
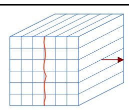
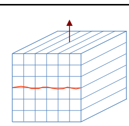
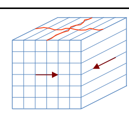
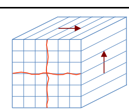
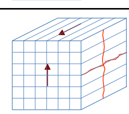
UD DAMAGE TYPES		**FIBER DAMAGE MODES**			**MATRIX DAMAGE MODES**		
UD DAMAGE MODES		**MODE 1uf j = 1**	**MODE 2uf j = 2**	**MODE 3uf j = 3**	**MODE 4um j = 4**	**MODE 5um j = 5**	**MODE 6um j = 6**
MODULI	q_{ij}^{U}						
E_a		**1**	**1**	**1**	**0**	**0**	**0**
E_b		**0**	**0**	**1**	**1**	**1**	**0**
E_c		**0**	**0**	**1**	**0**	**0**	**1**
G_{ab}		**1**	**1**	**1**	**1**	**1**	**0**
G_{bc}		**0**	**0**	**1**	**1**	**1**	**1**
G_{ca}		**1**	**1**	**1**	**0**	**0**	**1**

Figure 2: Coupling of Different Damage Modes to the Associated Reduction in Moduli for Unidirectional Lamina Model.

DAMAGE COUPLING MATRIX FOR FABRIC LAMINA MODEL

Eq. (22) is the damage coupling matrix, and Fig. 3 illustrates how Eq. (22) is associated with the modulus reduction for the fabric lamina model.

$$q_{ij}^F = \begin{bmatrix} 1 & 0 & 1 & 0 & 1 & 0 & 0 \\ 0 & 1 & 0 & 1 & 1 & 0 & 0 \\ 0 & 0 & 0 & 0 & 1 & 0 & 1 \\ 1 & 1 & 1 & 1 & 1 & 1 & 0 \\ 0 & 1 & 0 & 1 & 1 & 0 & 1 \\ 1 & 0 & 1 & 0 & 1 & 0 & 1 \end{bmatrix} \quad i = 1, \ldots, 6;\; j = 7, \ldots, 13. \tag{22}$$

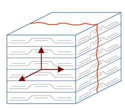
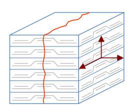
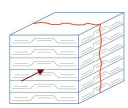
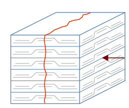
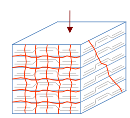
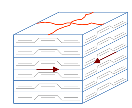
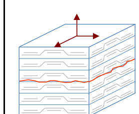
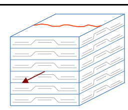
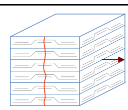
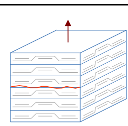
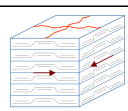
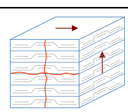
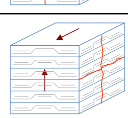
PW DAMAGE TYPES		**FIBER DAMAGE MODES**					**MATRIX DAMAGE MODES**	
PW DAMAGE MODES		**MODE 7f j = 7**	**MODE 8f j = 8**	**MODE 9f j = 9**	**MODE 10f j = 10**	**MODE 11f j = 11**	**MODE 12m j = 12**	**MODE 13m j = 13**
MO-DU-LI	q_{ij}^F							
$\mathbf{E_a}$		**1**	**0**	**1**	**0**	**1**	**0**	**0**
$\mathbf{E_b}$		**0**	**1**	**0**	**1**	**1**	**0**	**0**
$\mathbf{E_c}$		**0**	**0**	**0**	**0**	**1**	**0**	**1**
$\mathbf{G_{ab}}$		**1**	**1**	**1**	**1**	**1**	**1**	**0**
$\mathbf{G_{bc}}$		**0**	**1**	**0**	**1**	**1**	**0**	**1**
$\mathbf{G_{ca}}$		**1**	**0**	**1**	**0**	**1**	**0**	**1**

Figure 3: Coupling of Different Damage Modes to the Associated Reduction in Moduli for Fabric Lamina Model.

Through Eq. (15), the damage coupling matrix q_{ij} relates the individual damage variables ϖ_i to the various damage modes provided by the scalar damage functions ϕ_j for the unidirectional and fabric lamina models.

Unidirectional Fiber Modes 1uf, 2uf, & 3uf: For the unidirectional lamina model, the damage coupling vectors q_{i1} and q_{i2} of equation (21) are chosen such that the fiber tension-shear and compressive damage modes 1uf and 2uf, Eqs. (1) & (2), provide the reduction of elastic moduli E_a, G_{ab}, and G_{ca}, due to ϖ_1, ϖ_4 and ϖ_6 , respectively. The coupling vector q_{i3} provides that all the elastic moduli are reduced due to the fiber crush damage mode 3uf, Eq. (3).

Unidirectional Matrix Modes 3um, 4um, & 5um: For the transverse matrix damage modes 4um and 5um, Eqs. (4) & (5), q_{i4} and q_{i5} provide the reduction of E_b, G_{ab} and G_{bc}, while for the through thickness matrix damage mode 6um, q_{i6} provides the reduction of E_c, G_{bc}, and G_{ca}.

Fabric Fiber Modes 7f, 8f, 9f, 10f, & 11f: For the fabric lamina model, the damage coupling vectors q_{i7}, q_{i8}, q_{i9} and q_{i10} are chosen for the fiber tension-shear and compressive damage modes 7f to 10f, Eqs. (8), & (9); such that the fiber damage in either the fill or warp direction results in stiffness reduction in the loading direction and in the related shear directions. For the fiber crush damage mode 11f, Eq (10), the damage coupling vector q_{i11} is chosen such that all the stiffness values are reduced as an element is failed under the crush mode.

Fabric Matrix Modes 12m, & 13m: For the in-plane matrix shear failure mode 12m given by Eq. (11), the stiffness reduction due to q_{i12} is limited to in-plane shear modulus, while the through thickness matrix damage (delamination) mode 13m, the coupling vector q_{i13} is chosen for the through thickness tensile modulus and transverse shear moduli.

NON-LINEAR PROGRESSIVE DAMAGE MODEL OF MAT162

Utilizing the damage coupling matrix given by Eqs. (21) & (22), and the scalar damage function given by Eq. (20), the damage variables ϖ_i can be obtained from Eq. (15) for an individual failure mode j as:

$$\varpi_i = 1 - \exp\left\{\frac{1}{m_j}\left(1 - r_j^{m_j}\right)\right\} \qquad r_j \geq 1 \qquad (23)$$

Note that the damage thresholds r_j given in the damage criteria of Eqs. (1) to (12) are continuously increasing functions with increasing damage. The damage thresholds have an initial value of one, which results in a zero value for the associated damage variable ϖ_i from Eq. (23). This provides an initial elastic region bounded by the damage functions in strain space. The nonlinear response is modeled by loading on the damage surfaces to cause damage growth with increasing damage thresholds and the values of damage variables ϖ_i. After damage initiated, the progressive damage model assumes linear elastic response within the part of strain space bounded by the updated damage thresholds. The elastic response is governed by the reduced stiffness matrix associated with the updated damage variables ϖ_i given in Eq. (14).

In defining the non-linear stress-strain behavior of a composite material in a specific direction k, a damage threshold r_k (k = 1, …, 6) can also be expressed as the ratio between the current total strain in the kth direction and the corresponding yield strain.

$$r_k = \frac{\varepsilon_k}{\varepsilon_{ky}} \tag{24}$$

From Eq. 14, the Young's modulus in the kth direction can now be expressed as:

$$E_k = (1 - \varpi_k)E_{k0} = E_{k0}\exp\left\{\frac{1}{m_j}\left(1 - \left(\frac{\varepsilon_k}{\varepsilon_{ky}}\right)^{m_k}\right)\right\} \tag{25}$$

Since the reduced modulus is also considered linear, the stress-strain relationship of the damaged material can now be expressed as:

$$\sigma_k = E_k\varepsilon_k = E_{k0}\varepsilon_k\exp\left\{\frac{1}{m_j}\left(1 - \left(\frac{\varepsilon_k}{\varepsilon_{ky}}\right)^{m_k}\right)\right\} \tag{26}$$

Recognizing the fact that $\sigma_{ky} = E_{k0}\varepsilon_{ky}$ Eq. (26) can also be expressed as:

$$\frac{\sigma_k}{\sigma_{ky}} = \frac{\varepsilon_k}{\varepsilon_{ky}}\exp\left\{\frac{1}{m_j}\left(1 - \left(\frac{\varepsilon_k}{\varepsilon_{ky}}\right)^{m_k}\right)\right\} \tag{27}$$

Fig. 4 shows the plot of Eq. (27) for different values of damage softening parameter, *m*. Note that the value of $r = \varepsilon / \varepsilon_y \le 1$, represent the linear-elastic part of the stress-strain behavior, and Eq. (27) represents the post-yield damage softening behavior for $r = \varepsilon / \varepsilon_y \ge 1$.

It is well known that it is difficult to obtain the softening response of most quasi-brittle materials including fiber-reinforced composites. The softening response heavily depends on the set-up and test machines, which can lead to very scattered results. Consequently the choice of damage parameters for each mode becomes an open issue. Generally, smaller values of *m* make the material more ductile whereas higher values give the material more brittle behavior. A methodology to systematically determine the model material properties for penetration modeling has been successfully established in [Xiao et al., 2005].

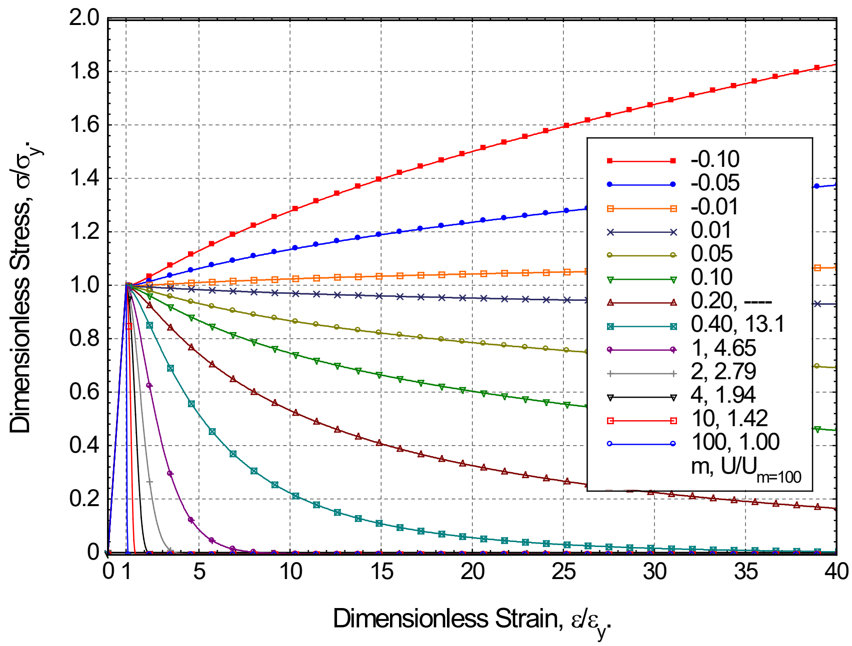


Figure 4: Non-Linear Progressive Damage Model of MAT162. Post-Yield Damage Softening of a Composite as a Function of Damage Softening Parameter *m*.

In MAT 162, the damage softening parameter m_1 controls the tensile and compressive fiber failure mode in a direction, and m_2 controls the transverse compressive matrix failure mode in b direction for the "unidirectional lamina model." However, for fabric the "fabric lamina model," m_2 controls the softening of tensile and compressive fiber failure mode in b direction. m_3 is for softening related to fiber crush mode, and m_4 is for both perpendicular and parallel matrix mode for "unidirectional" case, and for both in-plane matrix failure and through the thickness matrix failure for "fabric" case. Detail analysis on m parameters on the stress-strain behavior can be found in Ref. [Gama et al., 2009].

ADDITIONAL DISCUSSION ON MODULII AND STRENGTH REDUCTION

When fiber tension-shear damage is predicted in a layer by equation (1) or (8), the load carrying capacity of that layer in the associated direction is reduced to zero according to damage variable Eq. (23). For compressive fiber damage due to equation (2) or (9), the layer is assumed to carry a residual axial load in the damaged direction. The damage variables of Eq. (23) for the compressive modes have been modified to account for the residual strengths of $S_{aCR} = S_{aC} \times S_{FFC}$ and $S_{bCR} = S_{bC} \times S_{FFC}$ in the fill and warp directions, respectively.

For through thickness matrix (delamination) failure given by equation (6) or (12), the in-plane load carrying capacity within the element is assumed to be elastic (i.e., no in-plane damage). The load carrying behavior in the through thickness direction is assumed to depend on the opening or closing of the matrix damage surface. For tensile mode, $\varepsilon_c > 0$, the through thickness stress components are softened and reduced to zero due to the damage criteria described above. For compressive mode, $\varepsilon_c < 0$, the damage surface is considered to be closed, and thus, ε_c is assumed to be elastic, while ε_{bc} and ε_{ca} are allowed to reduce to sliding friction traction of equation (7) or (13). Accordingly, for the through thickness matrix failure under compressive mode, the damage variable equation is further modified such that the residual sliding strength value is equal to S_{SRC}.

EFFECT OF STRAIN RATES ON STRENGTH AND MODULI

The effect of strain-rate on the nonlinear stress-strain response of a composite layer is modeled by a logarithmic strain-rate dependent function for the moduli and strength of the form:

$$\frac{X_{RT}}{X_0} = 1 + C_{rate} \ln\left(\frac{\dot{\bar{\varepsilon}}}{\dot{\bar{\varepsilon}}_0}\right) \tag{28}$$

where, X_{RT} is the rate dependent property of interest at an average strain rate of $\dot{\bar{\varepsilon}}$, and X_0 is the quasi-static property of interest at an average reference strain rate of $\dot{\bar{\varepsilon}}_0$. In the present MAT162 formulation, the reference strain rate is chosen to be:

$$\dot{\bar{\varepsilon}}_0 = 1 \text{ per time unit} \tag{29}$$

If the time unit is sec, the reference strain rate is $1/s$. Similarly, if the time unit is μs, the the reference strain is $1/\mu s$.

EFFECT OF STRAIN RATE ON STRENGTH PROPERTIES

One average rate parameter, CERATE1 or C_{rate1} is used to add rate effects on strength properties as follows:

$$\{S_{RT}\} = \{S_0\}\left(1 + C_{rate1} \ln\left(\frac{\{\dot{\bar{\varepsilon}}\}}{\dot{\bar{\varepsilon}}_0}\right)\right) \tag{30}$$

where, the strength and strain rate matrices are given by Eq. (31). Note that the through thickness tensile strength S_{cT}, and the shear strengths S_{ab}, S_{bc}, & S_{ca}; are not considered as rate dependent in MAT162 formulation.

EFFECT OF STRAIN RATE ON MODULI

Three rate parameters, C_{rate2}, C_{rate3}, & C_{rate4} are used to add rate effects on three axial and three shear moduli as presented in Eq. (32), where, the moduli, strain rate, and rate parameter matrices are given by Eq. (33). Note that the rate effects on both the axial moduli, E_a & E_b, are controlled by the rate parameter C_{rate2}, and that for the through thickness modulus, E_c, by C_{rate4}. In addition, the rate effects on the shear moduli, G_{ab}, G_{bc}, & G_{ca}, are controlled by the rate parameter C_{rate3}.

$$\{S\} = \begin{Bmatrix} S_{aT} \\ S_{aC} \\ S_{bT} \\ S_{bC} \\ S_{FC} \\ S_{FS} \end{Bmatrix}, \ \& \ \{\dot{\bar{\varepsilon}}\} = \begin{Bmatrix} |\dot{\varepsilon}_a| \\ |\dot{\varepsilon}_a| \\ |\dot{\varepsilon}_b| \\ |\dot{\varepsilon}_b| \\ |\dot{\varepsilon}_c| \\ \left(\dot{\varepsilon}_{ca}^2 + \dot{\varepsilon}_{bc}^2\right)^{1/2} \end{Bmatrix} \tag{31}$$

$$\{E_{RT}\} = \{E_0\}\left(1 + \{C_{rate}\}\ln\frac{\{\dot{\bar{\varepsilon}}\}}{\dot{\varepsilon}_0}\right) \tag{32}$$

$$\{E_{RT}\} = \begin{Bmatrix} E_a \\ E_b \\ E_c \\ G_{ab} \\ G_{bc} \\ G_{ca} \end{Bmatrix}, \ \{\dot{\bar{\varepsilon}}\} = \begin{Bmatrix} |\dot{\varepsilon}_a| \\ |\dot{\varepsilon}_b| \\ |\dot{\varepsilon}_c| \\ |\dot{\varepsilon}_{ab}| \\ |\dot{\varepsilon}_{bc}| \\ |\dot{\varepsilon}_{ca}| \end{Bmatrix}, \ \& \ \{C_{rate}\} = \begin{Bmatrix} C_{rate2} \\ C_{rate2} \\ C_{rate4} \\ C_{rate3} \\ C_{rate3} \\ C_{rate3} \end{Bmatrix} \tag{33}$$

A discussion of rate dependent MAT162 properties can be found in Ref. [Gama & Gillespie Jr. 2011].

ELEMENT EROSION

A failed element is eroded in any of three different ways:

1. If fiber tensile failure in a "unidirectional" layer is predicted in the element and the axial tensile strain is greater than E_LIMIT. For a "fabric" layer, both in-plane directions are failed and exceed E_LIMIT.
2. If compressive relative volume (ratio of current volume to initial volume) in a failed element is smaller than ECRSH.
3. If expansive relative volume in a failed element is greater than EEXPN.

FINITE ELEMENT MODELING TIPS

- One point integration solid element (TYPE = 1) can be used for MAT162.
- In order to observe the delamination at the interface between two adjacent laminas, two different PART IDs with different MAT IDs for each parts and with different material orientation angles (BETA in the MAT162 cards) must be defined at the interface of interest. If it is required to model the delamination at the interface between two plies with the same material orientation angles, those two angles must be defined in different ways in each PART, e.g., BETA = 0.00 & BETA = 180.00.
- *DATABASE_EXTENT_BINARY must be included to check history variables.
- Type of *HOURGLASS need to be checked for minimum hourglass energy over the duration of the LS-DYNA solution.

DAMAGE HISTORY PARAMETERS

Information about the damage history variables for the associated failure modes can be plotted in LS-POST. These additional variables are tabulated below:

<table>
<tr><th colspan="3">History Variable</th><th rowspan="2">Description</th><th rowspan="2">Value</th><th rowspan="2">LS-POST Components</th></tr>
<tr><th>#</th><th>Uni</th><th>Fabric</th></tr>
<tr><td>1</td><td>max (r_1, r_2)</td><td>Max (r_7, r_9)</td><td>Fiber mode in a</td><td rowspan="5">0 - elastic
> 1- damage thresholds, Eqs. (1-6) to (8-12)</td><td>7</td></tr>
<tr><td>2</td><td>-</td><td>Max (r_8, r_{10})</td><td>Fiber mode in b</td><td>8</td></tr>
<tr><td>3</td><td>r_3</td><td>r_{11}</td><td>Fiber crush mode</td><td>9</td></tr>
<tr><td>4</td><td>r_5</td><td>r_{12}</td><td>Perpendicular matrix mode</td><td>10</td></tr>
<tr><td>5</td><td>r_6</td><td>r_{13}</td><td>Parallel matrix/ delamination mode</td><td>11</td></tr>
<tr><td colspan="3">6</td><td>Element delamination indicator</td><td>0 – no delamination
1 – with delamination</td><td>12</td></tr>
</table>

In LS-PREPOST, history variables of an element can be plotted from [History] tab. History var#7 to history var#12 are used for MAT162 damage variables. History variables can also be viewed as fringe component tab [FrinComp]. It is important to know that, in visualizing the MAT162 damage modes, the fringe range in the [FriRang] tab need to set to [User], the minimum and maximum value of the [User] range shall be set to [Min: = 0] & [Max: = 1] and finally the [Update] & [Done] buttons need to be pressed sequentially. The table above presents the relationship between damage thresholds, r_j, with the history variables for both "Unidirectional" and "Fabric" lamina models, and are presented separately in the following tables.

Unidirectional Lamina Damage Modes and Corresponding History Variables

Damage Mode	Mode No.	Equation	Eq. No.	Mat. Dir.	LS-PrePost HISV No.	HISV Values
Tension-Shear Fiber Mode	1uf	$f_1 - r_1^2 = \left(\frac{E_a\langle\varepsilon_a\rangle}{S_{aT}}\right)^2 + \left(\frac{G_{ab}^2\varepsilon_{ab}^2 + G_{ca}^2\varepsilon_{ca}^2}{S_{FS}^2}\right) - r_1^2 = 0$	(1)	1	7 [max{r_1, r_2}]	*
Compression Fiber Mode	2uf	$f_2 - r_2^2 = \left(\frac{E_a\langle\varepsilon_a'\rangle}{S_{aC}}\right)^2 - r_2^2 = 0$	(2)	1	7 [max{r_1, r_2}]	*
Crush Fiber Mode 3	3uf	$f_3 - r_3^2 = \left(\frac{E_c\langle-\varepsilon_c\rangle}{S_{FC}}\right)^2 - r_3^2 = 0$	(3)	3	9 [r_3]	*
Transverse Compression Matrix Mode	4um	$f_4 - r_4^2 = \left(\frac{E_b\langle-\varepsilon_b\rangle}{S_{bC}}\right)^2 - r_4^2 = 0$	(4)	2	8 [r_4]	*
Perpendicular Matrix Shear Mode	5um	$f_5 - r_5^2 = \left(\frac{E_b\langle\varepsilon_b\rangle}{S_{bT}}\right)^2 + \left(\frac{G_{bc}\varepsilon_{bc}}{S_{bc0}+S_{SRB}}\right)^2 + \left(\frac{G_{ab}\varepsilon_{ab}}{S_{ab0}+S_{SRB}}\right)^2 - r_5^2 = 0$	(5)	12	10 [r_5]	*
Parallel Matrix Shear Delamination Mode	6um	$f_6 - r_6^2 = S^2\left\{\left(\frac{E_c\langle\varepsilon_c\rangle}{S_{cT}}\right)^2 + \left(\frac{G_{bc}\varepsilon_{bc}}{S_{bc0}+S_{SRC}}\right)^2 + \left(\frac{G_{ca}\varepsilon_{ca}}{S_{ca0}+S_{SRC}}\right)^2\right\} - r_6^2 = 0$	(6)	13	11 [r_6]	*
Delamination Index					12 [r_6]	⁂

*	0 – elastic	> 1 – damage threshold [Equations (1) to (6)]
⁂	0 – no delamination	1 – full delamination

For the unidirectional damage modes, the damage coupling matrix [also presented below] given by Eq. (21) shows that for one specific damage mode, several elastic constants will be degraded.

- Tension-Shear Fiber Mode 1uf [TSFM-1uf]: $[E_A, G_{AB}, G_{CA}]$ will be degraded
- Compression Fiber Mode 2uf [CompFM-2uf]: $[E_A, G_{AB}, G_{CA}]$ will be degraded
- Crush Fiber Mode 3uf [CrshFM-3uf]: $[E_A, E_B, E_C, G_{AB}, G_{BC}, G_{CA}]$ will be degraded
- Trans. Comp. Matrix Mode 4um [TCMM-4um]: $[E_B, G_{AB}, G_{BC}]$ will be degraded
- Para. Matrix Shear Mode 5um [PMSM-5um]: $[E_B, G_{AB}, G_{BC}]$ will be degraded
- Perp. Matrix Shear Delam. Mode 6um [PMSDM-6um]: $[E_C, G_{BC}, G_{CA}]$ will be degraded

q_{ij}^U	TSFM-1uf	CompFM-2uf	CrshFM-3uf	TCMM-4um	PMSM-5um	PMSDM-6um
E_A	1	1	1	0	0	0
E_B	0	0	1	1	1	0
E_C	0	0	1	0	0	1
G_{AB}	1	1	1	1	1	0
G_{BC}	0	0	1	1	1	1
G_{CA}	1	1	1	0	0	1

Thus, for unidirectional lamina model, each damage mode will degrade three (3) moduli except for crush (6 moduli) mode. Hence loading in one direction may show more than one damage history variable activated.

Fabric Lamina Damage Modes and Corresponding History Variables

Damage Mode	Mode No.	Equation	Eq. No.	Mat. Dir.	LS-PrePost HISV No.	HISV Values
Tension-Shear Fiber Mode 7	7f	$f_7 - r_7^2 = \left(\frac{E_a\langle\varepsilon_a\rangle}{S_{aT}}\right)^2 + \left(\frac{G_{ca}\varepsilon_{ca}}{S_{aFS}}\right)^2 - r_7^{\;2} = 0$	(8a)	1	7 [max$\{r_7, r_9\}$]	*
Tension-Shear Fiber Mode 8	8f	$f_8 - r_8^2 = \left(\frac{E_b\langle\varepsilon_b\rangle}{S_{bT}}\right)^2 + \left(\frac{G_{bc}\varepsilon_{bc}}{S_{bFS}}\right)^2 - r_8^{\;2} = 0$	(8b)	2	8 [max$\{r_8, r_{10}\}$]	*
Compression Fiber Mode 9	9f	$f_9 - r_9^2 = \left(\frac{E_a\langle\varepsilon_a^{\prime}\rangle}{S_{aC}}\right)^2 - r_9^{\;2} = 0$	(9a)	1	7 [max$\{r_7, r_9\}$]	*
Compression Fiber Mode 10	10f	$f_{10} - r_{10}^2 = \left(\frac{E_b\langle\varepsilon_b^{\prime}\rangle}{S_{bC}}\right)^2 - r_{10}^{\;2} = 0$	(9b)	2	8 [max$\{r_8, r_{10}\}$]	*
Crush Fiber Mode 11	11f	$f_{11} - r_{11}^2 = \left(\frac{E_c\langle -\varepsilon_c\rangle}{S_{FC}}\right)^2 - r_{11}^{\;2} = 0$	(10)	3	9 $[r_{11}]$	*
In-Plane Matrix Shear Mode 12	12m	$f_{12} - r_{12}^2 = \left(\frac{G_{ab}\varepsilon_{ab}}{S_{ab}}\right)^2 - r_{12}^{\;2} = 0$	(11)	12	10 $[r_{12}]$	*
Parallel Matrix Shear Delamination Mode 13	13m	$f_{13} - r_{13}^2 = S^2\left\{\left(\frac{E_c\langle\varepsilon_c\rangle}{S_{cT}}\right)^2 + \left(\frac{G_{bc}\varepsilon_{bc}}{S_{bc0}+S_{SRC}}\right)^2 + \left(\frac{G_{ca}\varepsilon_{ca}}{S_{ca0}+S_{SRC}}\right)^2\right\} - r_{13}^{\;2} = 0$	(12)	13 & 23	11 $[r_{13}]$	*
Delamination Index					12 $[r_{13}]$	⁂

*	0 – elastic	> 1 – damage threshold [Equations (1) to (6)]
⁂	0 – no delamination	1 – full delamination

For the fabric damage modes, the damage coupling matrix [also presented below] given by Eq. (22) shows that for one specific damage mode, several elastic constants will be degraded.

- Tension-Shear Fiber Mode 7f [TSFM-7f]: $[E_A, G_{AB}, G_{CA}]$ will be degraded
- Compression Fiber Mode 9f [CompFM-9f]: $[E_A, G_{AB}, G_{CA}]$ will be degraded
- Tension-Shear Fiber Mode 8f [TSFM-8f]: $[E_B, G_{AB}, G_{BC}]$ will be degraded
- Compression Fiber Mode 10f [CompFM-10f]: $[E_B, G_{AB}, G_{BC}]$ will be degraded
- Crush Fiber Mode 11f [CrshFM-11f]: $[E_A, E_B, E_C, G_{AB}, G_{BC}, G_{CA}]$ will be degraded
- In-Plane Matrix Shear Mode 12m [IPMSM-11m]: $[G_{AB}]$ will be degraded
- Perp. Matrix Shear Delam. Mode 12m [PMSDM-6um]: $[E_C, G_{BC}, G_{CA}]$ will be degraded

$\mathbf{q_{ij}^U}$	TSFM-7f	TSFM-8f	CompFM-9f	CompFM-10f	CrshFM-11f	IPMSM-12m	PMSDM-13m
$\mathbf{E_A}$	1	0	1	0	1	0	0
$\mathbf{E_B}$	0	1	0	1	1	0	0
$\mathbf{E_C}$	0	0	0	0	1	0	1
$\mathbf{G_{AB}}$	1	1	1	1	1	0	0
$\mathbf{G_{BC}}$	0	1	0	1	1	1	1
$\mathbf{G_{CA}}$	1	0	1	0	1	0	1

Thus each damage mode will degrade three (3) moduli except for crush (6 moduli) and in-plane shear (1 modulus). Hence loading in one direction may show more than one damage history variable activated.

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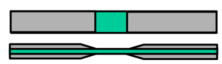
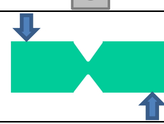
APPENDIX A: MAT162 PROPERTIES AND PARAMETERS

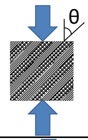
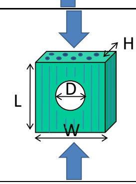
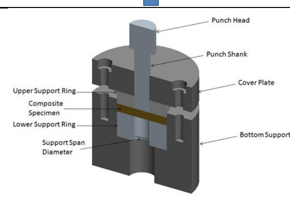
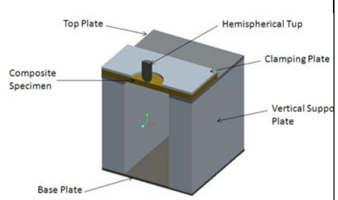
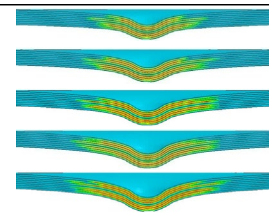
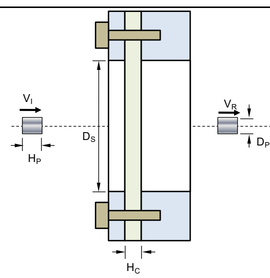
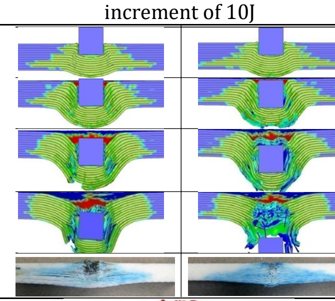
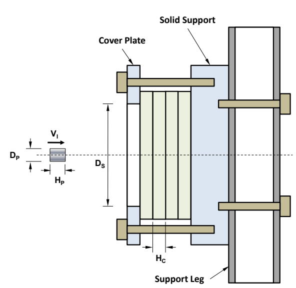
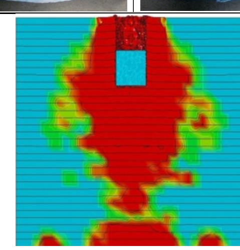
MAT162 ELASTIC AND STRENGTH PROPERTIES

In addition to ASTM standard test methods, UD-CCM has developed non-standard experimental techniques and computational methodologies to determine all material properties and parameters needed for MAT162.

MAT162 ELASTIC & STRENGTH PROPERTIES: ASTM STANDARD TESTS

Properties, Parameters	Test Method	Specimen	Dimensions (mm)	Miscellaneous
E_1, ν_{12}, X_1^T	0° Tension (ASTM D3039)		254×25.4×1	
E_2, ν_{21}, X_2^T	90° Tension (ASTM D3039)		175×25.4×2	
X_1^C	0° Compression (ASTM D3410)		155×25.4×2	
X_2^C	90° Compression (ASTM D3410)		155×25.4×2	
$E_3, \nu_{31}, \nu_{32}, X_3^T$	Thru-thickness Tension/ Compression (no standard)		20×20×20 15×15×30	
G_{12}, G_{23}, G_{31} S_{12}, S_{23}, S_{31}	In-Plane +/-45 Tension, Rail-Shear & V-Notch Shear (ASTM D5379)		76×4.5×20	Shear in 1-2, 2-3, 3-1 planes

MAT162 PROPERTIES & PARAMETERS: NON-STANDARD UD-CCM TEST & COMPUTATIONAL METHODOLOGIES

$E_3, PHIC$	Out-of-Plane Off-Axis Compression (no standard)		15×15×15	θ = 0°, 15°, 30°, 45°, 60°, 75°
$SFFC$	Open Hole Compact Compression Test (no standard)		25×25×13 D = 8~12	Loading along fiber direction for both UD and PW Composites
SFC, SFS	Quasi-Static Punch Shear Test (QS-PST) Punch Crush Strength (PCS) & Punch Shear Strength (PSS) (BZH Methodology)		25 mm Discs 100x100 Plates 150x150 Plates	SPR = D_S/D_P SPR = 0 for SFC SPR = 1.1 for SFS
$OMGMX$ & Damage Softening Parameters m_1 to m_4	Low Velocity Impact Experiments; & Numerical Simulation (BZH Methodology)		100x150 Plates	 Energy Levels: 30J to 70J @ an increment of 10J
Erosion Parameters E_LIMT $EEXPN$ Rate Parameters C_{rate1} to C_{rate4}	Hopkinson Bar Testing Dynamic Punch Shear Ballistic Testing Numerical Simulation of Ballistic Tests (BZH Methodology)		150x150x15 Plates	
Crush Parameters, $SFC, ECRSH$	Depth of Penetration Impact Experiments; & Numerical Simulation (BZH Methodology)		305x205x50 Plates	 Impact at 300 to 800 m/s @ an interval of 50 m/s. Measure DoP as a function of impact velocity. Parametrically determine SFC & ECRSH to match the Experimental Data.

MAT162 DATABASE OF COMPOSITE PROPERTIES

Properties, Unit	UD S-2 Glass/SC15	PW S-2 Glass/SC15	PW S-Glass/Phenolic OWENS Corning	PW E-Glass/Phenolic U.S. AERDC
V_f	0.60	0.53	0.62	0.66
ρ_C, g/cm3	1.85	1.85	2.00	2.107
E1, GPa	64.0	27.5	38.6	29.15
E2, GPa	11.8	27.5	31.9	29.15
E3, GPa	11.8	11.8	12.0	11.00
ν21	0.0535	0.110	0.100	0.078
ν31	0.0535	0.180	0.200	0.109
ν32	0.449	0.180	0.200	0.109
G12, GPa	4.30	2.90	4.50	1.54
G23, GPa	3.70	2.14	2.90	1.67
G31, GPa	4.30	2.14	3.10	1.67
X1T, MPa	1380	604	402	531
X1C, MPa	770	291	138	131
X2T, MPa	47	604	592	531
X2C, MPa	137	291	204	131
X3T, MPa	47	58	52	50
SFC, MPa	850	850	1540	870
SFS, Mpa	250	300	172	160
S12, MPa	76	75	73	35
S23, MPa	38	58	49	27
S31, MPa	76	58	49	27
AM1	100.00	2.00	1.00	1.00
AM2	10.00	2.00	1.00	1.00
AM3	1.00	0.50	0.50	0.50
AM4	0.10	0.20	-0.20	0.20/-0.20
PHIC	10	10	10	10
SFFC	0.10	0.30	0.30	0.30
Crate1	0.00	0.03	0.03	0.03
Crate2	0.00	0.00	0.00	0.00
Crate3	0.00	0.03	0.03	0.03
Crate4	0.00	0.03	0.03	0.03
SDELM	1.2	1.2	1.2	1.2
OMGMX	0.999	0.999	0.998 (LVI) 0.997 (DoP	0.994
E_LIMT	4.5	4.5	5.0	4.0
EEXPN	4.5	4.5	5.0	4.0
ECRSH	0.001	0.001	0.700	0.500
AMODEL	1 (UD)	2 (PW)	2 (PW)	2 (PW)
SOURCE	**Ref. [A]**	**Ref. [B]**	**Ref. [C]**	**Ref. [D]**

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APPENDIX B: GENERAL DISCUSSION

DISCUSSION ON REFERENCE STRAIN RATE

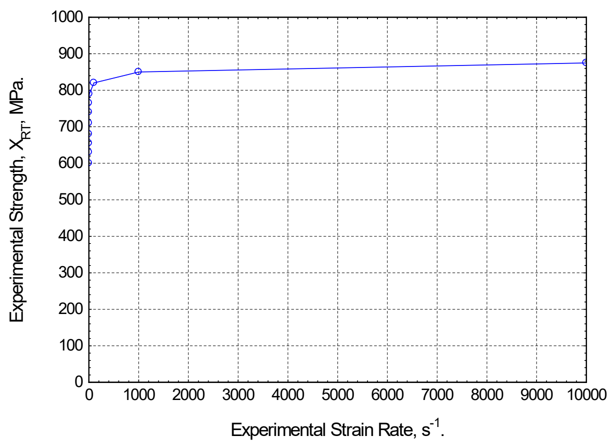
The reference strain rate in MAT162 is set to 1. If the time unit in LS-DYNA computation is set to seconds (s), then this reference strain rate is $\dot{\bar{\varepsilon}}_0 = 1s^{-1}$. This raises a question of how to calculate the rate parameters, C_{rate}, for different time units of LS-DYNA computations.

Consider a fictitious experimental set of strength data as a function of strain rates measured in s^{-1}. Table SR-1 shows this set of data. Fig. SR-1a shows the plot of this set of experimental data. The MAT162 rate equation is expressed as:

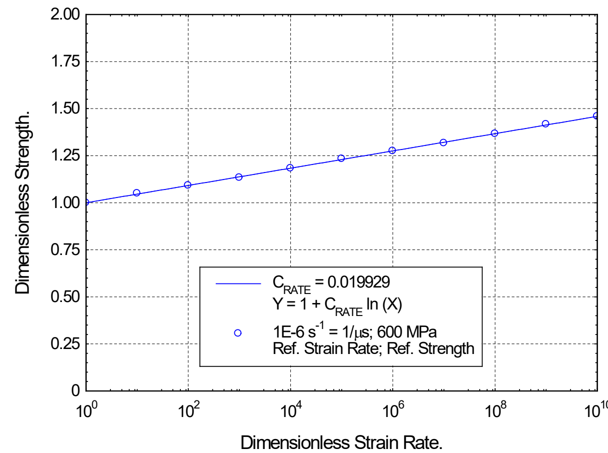
$$\frac{X_{RT}}{X_0} = 1 + C_{rate} \ln\left(\frac{\dot{\bar{\varepsilon}}}{\dot{\bar{\varepsilon}}_0}\right) \quad \text{(SR-1)}$$

Table SR-1: A Fictitious Experimental Set of Strength Data and Dimensionless Data for Reference Strain Rate, $\dot{\bar{\varepsilon}}_0 = 1.0 \times 10^{-6}\ s^{-1}$.

Strain Rate, s^{-1}	Strength, MPa	$\dot{\bar{\varepsilon}}_0 = 1.0 \times 10^{-6}\ s^{-1}$	$X_0 = 600\ MPa$
$\dot{\bar{\varepsilon}}$	X_{RT}	$\dot{\bar{\varepsilon}}/\dot{\bar{\varepsilon}}_0$	X_{RT}/X_0
1.00E-06	600	1.00E+00	1.0000
1.00E-05	630	1.00E+01	1.0500
1.00E-04	655	1.00E+02	1.0917
1.00E-03	680	1.00E+03	1.1333
1.00E-02	710	1.00E+04	1.1833
1.00E-01	740	1.00E+05	1.2333
1.00E+00	765	1.00E+06	1.2750
1.00E+01	790	1.00E+07	1.3167
1.00E+02	820	1.00E+08	1.3667
1.00E+03	850	1.00E+09	1.4167
1.00E+04	875	1.00E+10	1.4583



(a) Experimental Data



(b) MAT162 Rate Equation for Ref. Strain Rate, $\dot{\varepsilon}_0 = 1.0 \times 10^{-6}\ s^{-1}$

Figure SR-1: A Fictitious Experimental Set of Strength MAT162 Rate Equation for Reference Strain Rate, $\dot{\varepsilon}_0 = 1.0 \times 10^{-6}\ s^{-1}$.

We will use the Table SR-1 data to determine the rate parameter for different time units in LS-DYNA computations.

1. Reference Strain Rate $\dot{\boldsymbol{\varepsilon}}_0 = \mathbf{1.0} \times \mathbf{10^{-6}}\ \boldsymbol{s^{-1}} = \mathbf{1}/\boldsymbol{\mu s}$ for LS-DYNA Time Unit of Micro-Second

Consider the LS-DYNA time unit be micro-second. Also consider the reference strain rate in micro-second time unit to be $\dot{\varepsilon}_0 = 1\ \mu s^{-1} = 1.0 \times 10^{-6}\ s^{-1}$. From Table SR-1, the reference strength is $X_0 = 600\ MPa$. We can then normalize the experimental data with the reference strain rate and reference strength and the dimensionless values $(\dot{\varepsilon}/\dot{\varepsilon}_0, X_{RT}/X_0)$ are also presented in Table SR-1. The dimensionless strength and strain rates can then be plotted and is presented in Fig. SR-1b. Eq. (SR-1) can then be used to fit the dimensionless experimental data in determining the rate parameter, C_{rate}, and for the fictitious experimental data presented in Table SR-1 is found to be, $[C_{rate}]_{\dot{\varepsilon}_0 = 1\ \mu s^{-1}} = 0.019929$.

2. Reference Strain Rate $\dot{\boldsymbol{\varepsilon}}_0 = \mathbf{1.0} \times \mathbf{10^{-3}}\ \boldsymbol{s^{-1}} = \mathbf{1}/\boldsymbol{ms}$ for LS-DYNA Time Unit of Milli-Second

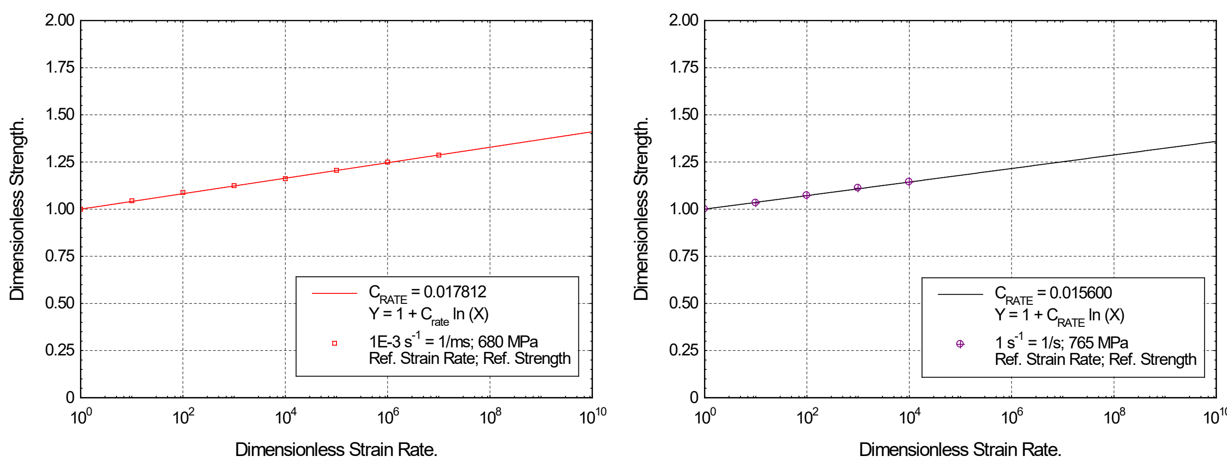
Consider the LS-DYNA time unit be milli-second. Also consider the reference strain rate in milli-second time unit to be $\dot{\varepsilon}_0 = 1\ ms^{-1} = 1.0 \times 10^{-3}\ s^{-1}$. The corresponding reference strength is $X_0 = 680\ MPa$. Table SR-2 shows the dimensionless strain rate and stress data and is plotted in Fig. SR-2a. Note that strain rates $< 1.0 \times 10^{-3}\ s^{-1}$ are not considered. This data is curve fitted to determine the rate parameter and is found to be, $[C_{rate}]_{\dot{\varepsilon}_0 = 1\ ms^{-1}} = 0.017812$.

Table SR-2: A Fictitious Experimental Set of Strength Data and Dimensionless Data for Reference Strain Rate, $\dot{\varepsilon}_0 = 1.0 \times 10^{-3}\ s^{-1}$, and $\dot{\varepsilon}_0 = 1.0\ s^{-1}$.

Strain Rate, s^{-1}	Strength, MPa	$\dot{\varepsilon}_0 = 1.0 \times 10^{-3}\ s^{-1}$	$X_0 = 680\ MPa$	$\dot{\varepsilon}_0 = 1.0\ s^{-1}$	$X_0 = 765\ MPa$
$\dot{\varepsilon}$	X_{RT}	$\dot{\varepsilon}/\dot{\varepsilon}_0$	X_{RT}/X_0	$\dot{\varepsilon}/\dot{\varepsilon}_0$	X_{RT}/X_0
1.00E-06	600	-	-	-	-
1.00E-05	630	-	-	-	-
1.00E-04	655	-	-	-	-
1.00E-03	680	1.00E+00	1.0000	-	-
1.00E-02	710	1.00E+01	1.0441	-	-
1.00E-01	740	1.00E+02	1.0882	-	-
1.00E+00	765	1.00E+03	1.1250	1.00E+00	1.0000
1.00E+01	790	1.00E+04	1.1618	1.00E+01	1.0327
1.00E+02	820	1.00E+05	1.2059	1.00E+02	1.0719
1.00E+03	850	1.00E+06	1.2500	1.00E+03	1.1111
1.00E+04	875	1.00E+07	1.2868	1.00E+04	1.1438

3. *Reference Strain Rate $\dot{\varepsilon}_0 = 1.0\ s^{-1} = 1/s$ for LS-DYNA Time Unit of Second*

Consider the LS-DYNA time unit be second. Also consider the reference strain rate in second time unit to be $\dot{\varepsilon}_0 = 1\ s^{-1}$. The corresponding reference strength is $X_0 = 765\ MPa$. Table SR-2 shows the dimensionless strain rate and stress data and is plotted in Fig. SR-2b. Note that strain rates $< 1.0\ s^{-1}$ are not considered. This data is curve fitted to determine the rate parameter and is found to be, $[C_{rate}]_{\dot{\varepsilon}_0 = 1\ ms^{-1}} = 0.015600$.



(a) MAT162 Rate Equation for Ref. Strain Rate, $\dot{\varepsilon}_0 = 1.0 \times 10^{-3}\ s^{-1}$

(b) MAT162 Rate Equation for Ref. Strain Rate, $\dot{\varepsilon}_0 = 1.0\ s^{-1}$

Figure SR-2: MAT162 Rate Equations for Reference Strain Rates, $\dot{\varepsilon}_0 = 1.0 \times 10^{-3}\ s^{-1}$, and $\dot{\varepsilon}_0 = 1.0\ s^{-1}$.

DISCUSSION ON LAMIANTE ARCHITECTURE AND PREDEFINED DELAMINATION PLANES

A composite laminate may contain several numbers of laminas or plies stacked through-the-thickness of the laminate. If the individual laminas are very thin, it is suggested to combine several laminas into one sub-laminate. Figure LA-1 shows the finite element model of such a sub-laminate with three (3) through-thickness elements.

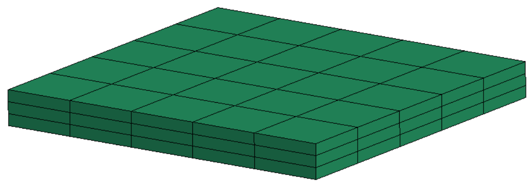


Figure LA-1: Finite Element Model of a Sub-Laminate with Three (3) Through-Thickness Elements.

Once a sub-laminate model is created, one should assign a Part ID to the sub-laminate and associate the **PID** with a Material ID with a pre-defined material angle (**BETA**), e.g.; **PID=1, MID=100, BETA=0** in Fig. LA-2. Several sub-laminates can be stacked through-the-thickness to build a laminate. Figure LA-2 shows a composite laminate with four (4) sub-laminates stacked through-thickness with the stacking sequence [0/90/0/90] and each sub-laminates are assigned with different Part IDs, i.e., **PID=1, 2, 3, 4**. However, since the stacking sequence is taken as [0/90/0/90], only two Material IDs (i.e., **MID=100, 200**) are sufficient. Note that all duplicate nodes between the PIDs 1 to 4 needs to be merged and the unreferenced nodes be deleted.

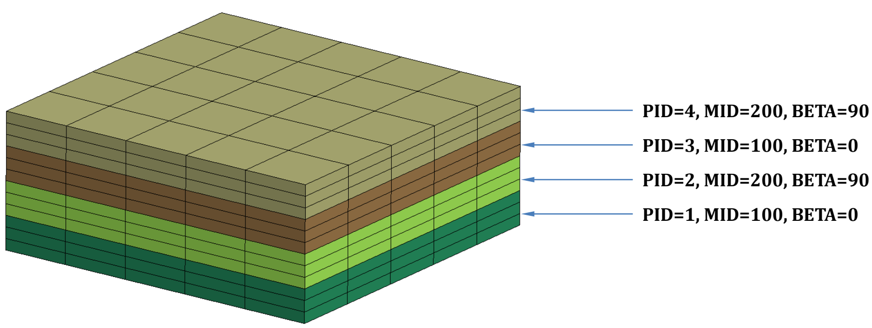


Figure LA-2: Node Merged Finite Element Model of a Laminate consisting of Four (4) Sub-Laminates with the Stacking Sequence [0/90/0/90]. Three Delamination Planes are thus Predefined between PIDs 1&2, 2&3, and 3&4.

According to MAT162 formulations, three pre-defined delamination planes will be automatically defined at the interface between four parts with different material angles (**BETA**). Once delamination between two parts with different material angles is predicted, shear properties of the elements adjacent to the delamination interface will be degraded to mimic delamination without creating physical surfaces between sub-laminates or parts. The advantage of MAT162 delamination criterion is that it is simple, however, the disadvantage is that the predicted delamination is assigned a thickness equal to one element near the delamination interface and is not physical in nature. This is why, three elements through-the-thickness of a sub-laminate or part is proposed so that the pseudo-delamination is limited to one-third of the thickness of a sub-laminate or part.

Figures LA-3 to 5 show three additional laminate stacking sequences defined with four sub-laminates.

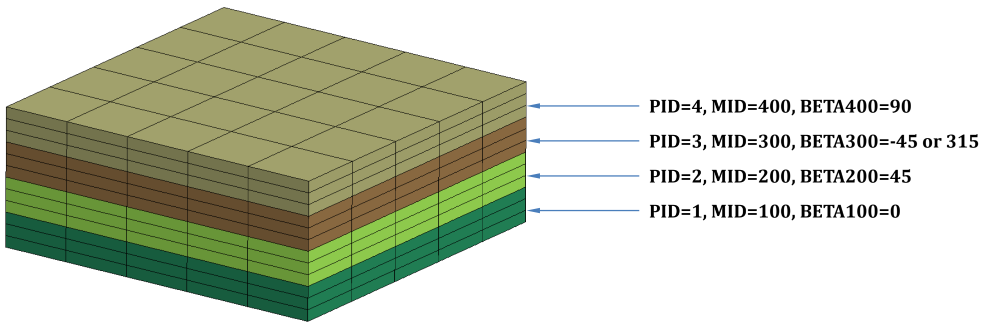


Figure LA-3: Node Merged Finite Element Model of a Laminate consisting of Four (4) Sub-Laminates with the Stacking Sequence [0/45/-45/90]. Three Delamination Planes are thus Predefined between PIDs 1&2, 2&3, and 3&4.

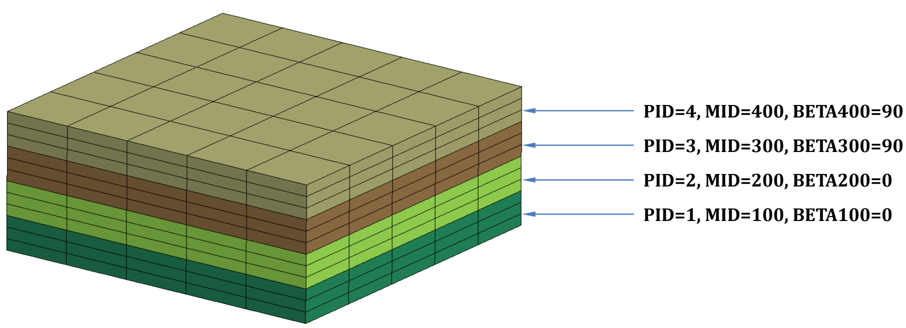


Figure LA-4: Node Merged Finite Element Model of a Laminate consisting of Four (4) Sub-Laminates with the Stacking Sequence [0/0/90/90]. One Delamination Plane is thus Predefined between PIDs 2&3.

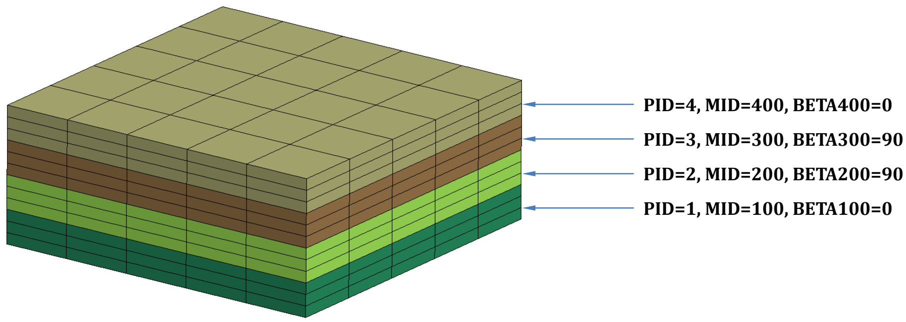


Figure LA-5: Node Merged Finite Element Model of a Laminate consisting of Four (4) Sub-Laminates with the Stacking Sequence [0/90/90/0]. Two Delamination Planes are thus Predefined between PIDs 1&2, and 3&4.

DISCUSSION ON CONTROL ACCURACY

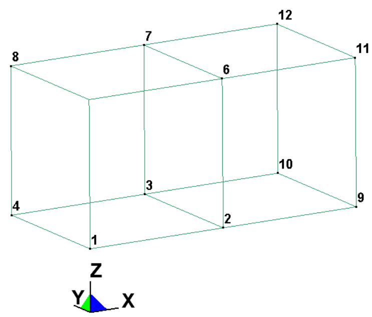


Figure CA-1: Definition of Two Single Elements in the Discussion of Control Accuracy.

In Fig. CA-1, if we choose **AOPT=0** for a composite element and define the element (**EID=1**) with the following node sequence, then the vector **V12** connecting nodes 1 & 2 defines the material direction 1 or A. The vector **V14** connecting nodes 1 & 4 together with the vector **V12** defines the material plane 1-2 or A-B, and the vector cross product **V12** × **V14** defines the through-thickness material direction 3 or C.

```
$------+-------+-------+-------+-------+-------+-------+-------+-------+-------+
*ELEMENT_SOLID
$------+-------+-------+-------+-------+-------+-------+-------+-------+-------+
$#   eid     pid      n1      n2      n3      n4      n5      n6      n7      n8
$------+-------+-------+-------+-------+-------+-------+-------+-------+-------+
       1     100       1       2       3       4       5       6       7       8
$------+-------+-------+-------+-------+-------+-------+-------+-------+-------+
```

Following the same procedure, one can define the second element (**EID=2**) as follows to be consistent with **AOPT=0**.

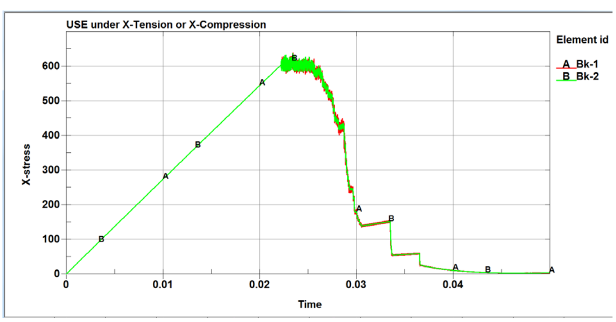
```
$------+-------+-------+-------+-------+-------+-------+-------+-------+-------+
*ELEMENT_SOLID
$------+-------+-------+-------+-------+-------+-------+-------+-------+-------+
$#   eid     pid      n1      n2      n3      n4      n5      n6      n7      n8
$------+-------+-------+-------+-------+-------+-------+-------+-------+-------+
       2     100       9      10       3       2      11      12       7       6
$------+-------+-------+-------+-------+-------+-------+-------+-------+-------+
```

However, if the second element (**EID=2**) is defined as follows, then it is not consistent with **AOPT=0**.

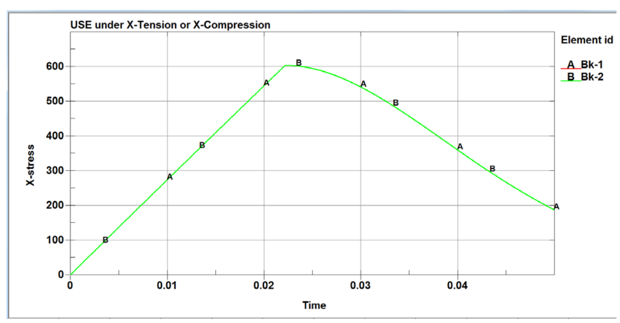
```
$------+-------+-------+-------+-------+-------+-------+-------+-------+-------+
*ELEMENT_SOLID
$------+-------+-------+-------+-------+-------+-------+-------+-------+-------+
$#   eid     pid      n1      n2      n3      n4      n5      n6      n7      n8
$------+-------+-------+-------+-------+-------+-------+-------+-------+-------+
       1     100       1       2       3       4       5       6       7       8
       2     100       2       9      10       3       6      11      12       7
$------+-------+-------+-------+-------+-------+-------+-------+-------+-------+
```

One can try to fix this problem by choosing **AOPT=2** by defining two vectors **A**(1,0,0) & **B**(0,1,0) to represent material direction 1 or A and the plane 1-2 or A-B, but this will not correct the problem, instead produce a wrong material response.

Figure CA-2a shows the wrong stress-time plot using **AOPT=2** with the wrong element definition for **EID=2**. In order to solve this problem, the LS-DYNA control card ***CONTROL_ACCURACY** can be used which allows **INVARIENT NODE NUMBERING** if the parameter **INN=3** is chosen for solid elements. Figure CA-2b shows the correct stress-time response with **INN=3** in the control accuracy card.



(a) AOPT=2 with INN=1



(a) AOPT=2 with INN=3

Figure CA-2: Definition of Two Single Elements in the Discussion of Control Accuracy.

EXAMPLE OF *CONTROL_ACCURACY CARD WITH PARAMETERS

```
$--------+---------+---------+---------+---------+---------+---------+---------+
*PARAMETER_EXPRESSION
$--------+---------+---------+---------+---------+---------+---------+---------+
$ PRMR1   EXPRESSION1
$--------+---------+---------+---------+---------+---------+---------+---------+
I INN     3
$--------+---------+---------+---------+---------+---------+---------+---------+
...
...
...
...
...
$--------+---------+---------+---------+---------+---------+---------+---------+
*CONTROL_ACCURACY
$--------+---------+---------+---------+---------+---------+---------+---------+
$#     osu       inn    pidosu
         0      &INN
$--------+---------+---------+---------+---------+---------+---------+---------+
```

AXIS OPTION AOPT AND SWITCHING MATERIAL AXES

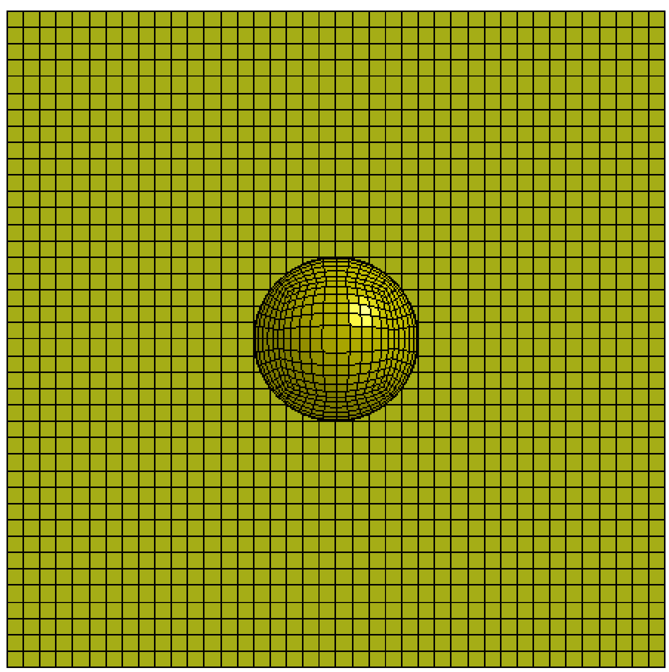
Axis options, **AOPT=0,2,4** has been tested and are found to be working with MAT162 along with the material axes switch options **MACF** without any problems. At present, we are looking at the applicability of axis option **AOPT=1**, and will report when our test is complete.

APPENDIX C: MAT162 EXAMPLES

EXAMPLE 1: SPHERE IMPACT ON COMPOSITES
An Example of MAT162 Progressive Composite Damage Model

A SIMPLE FEA MODEL OF A FOUR LAYER COMPOSITE LAMINATE WITH ONE DELAMINATION PLANE

The FEA model has two parts, each part defining two layers/laminas of a four layers composite laminate. The two bottom layers/laminas with PART ID = 101 is assigned a material angle of zero degree by defining the parameter BETA101 = 0.00 and the same for the top part with PART ID = 102 with the parameter BETA102 = 90.00. A delamination plane is thus defined between two different PART IDs each having a different material angle defined by BETA. For example, if the parameter BETA102 is changed to zero degree, there will be no delamination plane defined in the model.



The input files used in this example are summarized below.

FILE: 001-Sphere-Impact-on-Composite-Plate.key

```
$--------+---------+---------+---------+---------+--------+---------+---------+
$ Copyright 2015 (C) University of Delaware Center for Composite Materials
$--------+---------+---------+---------+---------+--------+---------+---------+
$ Unit System: mm-tonne-s (milimeter-tonne-second-Newton-MegaPascal)
$--------+---------+---------+---------+---------+--------+---------+---------+
$ Date: April 15, 2015
$ Dr. Bazle Z. (Gama) Haque
$ Senior Scientist of Center for Composite Materials (UD-CCM)
$ Assistant Professor of Mechanical Engineering
$ University of Delaware
$ Newark, DE 19716, USA
$ Tel: (302) 690-4741; E-mail: gama@udel.edu
$--------+---------+---------+---------+---------+--------+---------+---------+
$ Visualizing Damage Modes in LS-PrePost
$ Page-1 Tab-Range: User Min=0 Max=1
$ Toggle Switch: Lcon-ON
$ Page-1 Tab-Fcomp: Misc history var#7  (Fiber Tension-Shear Along A)
$ Page-1 Tab-Fcomp: Misc history var#8  (Fiber Tension-Shear Along B)
$ Page-1 Tab-Fcomp: Misc history var#9  (Fiber Crush)
$ Page-1 Tab-Fcomp: Misc history var#10 (In-Plane Matrix Crack)
$ Page-1 Tab-Fcomp: Misc history var#11 (Transverse Matrix Crack)
$ Page-1 Tab-Fcomp: Misc history var#12 (Delamination between 0 & 90 pliies)
$ Delamination between 0 & 90 pliies for this specific problem
$--------+---------+---------+---------+---------+--------+---------+---------+
$# Change the NCPU below to match your desired number of CPUs.
$--------+---------+---------+---------+---------+--------+---------+---------+
*KEYWORD MEMORY=10M NCPU=16
$--------+---------+---------+---------+---------+--------+---------+---------+
*PARAMETER_EXPRESSION
$--------+---------+---------+---------+---------+--------+---------+---------+
R us      1.00E-6
R ENDTIME 600.00*us
R NOD3    30.0
R DTBIN   0.01*us
R DTD3    ENDTIME/NOD3
R TSSFAC  0.950
$--------+---------+---------+---------+---------+--------+---------+---------+
I BSORT   1
I DEPTH   5
R PARMAX  1.0025
I SBOPT   5
R SLSFAC  0.010
R SLDTHK  0.001
I NEIPH   36
$--------+---------+---------+---------+---------+--------+---------+---------+
R mps     1000.00
R VX      +0.001*mps
R VY      +0.001*mps
R VZ      -100.00*mps
$--------+---------+---------+---------+---------+--------+---------+---------+
R mm      1.000
R dx      0.01*mm
R dy      0.01*mm
R dz      12.10*mm
$--------+---------+---------+---------+---------+--------+---------+---------+
$ *** MAT162 PROPERTIES & PARAMETERS ***
$--------+---------+---------+---------+---------+--------+---------+---------+
R gmpcc   1.0E-9
R MPa     1.000
R GPa     1.0E+3
$--------+---------+---------+---------+---------+--------+---------+---------+
I MIDP    700
R ROP     7.85*gmpcc
R EP      210.00*GPa
R PRP     0.290
$--------+---------+---------+---------+---------+--------+---------+---------+
I MID101  101
I MID102  102
R BETA101 0.00
```

```
R BETA102 90.00
$--------+---------+---------+---------+---------+---------+---------+---------+
R DENSITY 1.850*gmpcc
R E11     27.5*GPa
R E22     27.5*GPa
R E33     11.8*GPa
$--------+---------+---------+---------+---------+---------+---------+---------+
R NU21    0.110
R NU31    0.180
R NU32    0.180
$--------+---------+---------+---------+---------+---------+---------+---------+
R G12     2.90*GPa
R G23     2.14*GPa
R G31     2.14*GPa
$--------+---------+---------+---------+---------+---------+---------+---------+
R X1T     604.0*MPa
R X1C     291.0*MPa
R X2T     604.0*MPa
R X2C     291.0*MPa
R X3T     58.0*MPa
$--------+---------+---------+---------+---------+---------+---------+---------+
R SFC     850.0*MPa
R SFS     300.0*MPa
$--------+---------+---------+---------+---------+---------+---------+---------+
R S12     75.0*MPa
R S23     58.0*MPa
R S31     58.0*MPa
$--------+---------+---------+---------+---------+---------+---------+---------+
R SFFC    0.300
R PHIC    10.00
R SDELAM  1.200
$--------+---------+---------+---------+---------+---------+---------+---------+
R OMGMX   0.999
R ECRSH   0.001
R ELIMIT  4.000
R EEXPN   4.000
$--------+---------+---------+---------+---------+---------+---------+---------+
R AM1FD1  2.00
R AM2FD2  2.00
R AM3FCPS 0.50
R AM4MCDE 0.20
$--------+---------+---------+---------+---------+---------+---------+---------+
R CR1SIG  0.030
R CR2E1E2 0.000
R CR3G    0.030
R CR4E3   0.030
$--------+---------+---------+---------+---------+---------+---------+---------+
*MAT_ELASTIC_TITLE
$--------+---------+---------+---------+---------+---------+---------+---------+
Sphere Projectile
$--------+---------+---------+---------+---------+---------+---------+---------+
$#     mid        ro         e        pr        da        db  not used
     &MIDP      &ROP       &EP      &PRP
$--------+---------+---------+---------+---------+---------+---------+---------+
*MAT_COMPOSITE_DMG_MSC_TITLE
$--------+---------+---------+---------+---------+---------+---------+---------+
PW S-2 Glass/SC-15 0 Layer with beta = 0
$--------+---------+---------+---------+---------+---------+---------+---------+
$#     mid        ro        ea        eb        ec      prba      prca      prcb
   &MID101  &DENSITY      &E11      &E22      &E33     &NU21     &NU31     &NU32
$#     gab       gbc       gca      aopt      macf
      &G12      &G23      &G31       2.0         0
$#      xp        yp        zp        a1        a2        a3
     0.000     0.000     0.000     1.000     0.000     0.000
$#      v1        v2        v3        d1        d2        d3      beta
     0.000     0.000     0.000     0.000     1.000     0.000  &BETA101
$#     sat       sac       sbt       sbc       sct       sfc       sfs       sab
      &X1T      &X1C      &X2T      &X2C      &X3T      &SFC      &SFS      &S12
$#     sbc       sca      sffc    amodel      phic    e_limt    s_delm
      &S23      &S31     &SFFC       2.0     &PHIC   &ELIMIT   &SDELAM
```

```
$#   omgmx     ecrsh     eexpn   cr1/sig   am1/fda
    &OMGMX    &ECRSH    &EEXPN   &CR1SIG   &AM1FD1
$# am2/fdb am3/sfc/s am4/mc/de cr2/ea/eb     cr3/g    cr4/ec
   &AM2FD2  &AM3FCPS  &AM4MCDE  &CR2E1E2     &CR3G    &CR4E3
$--------+---------+---------+---------+---------+---------+---------+---------+
*MAT_COMPOSITE_DMG_MSC_TITLE
$--------+---------+---------+---------+---------+---------+---------+---------+
PW S-2 Glass/SC-15 0 Layer with beta = 90
$--------+---------+---------+---------+---------+---------+---------+---------+
$#     mid        ro        ea        eb        ec      prba      prca      prcb
   &MID102  &DENSITY      &E11      &E22      &E33     &NU21     &NU31     &NU32
$#     gab       gbc       gca      aopt      macf
      &G12      &G23      &G31       2.0         0
$#      xp        yp        zp        a1        a2        a3
     0.000     0.000     0.000     1.000     0.000     0.000
$#      v1        v2        v3        d1        d2        d3      beta
     0.000     0.000     0.000     0.000     1.000     0.000  &BETA102
$#     sat       sac       sbt       sbc       sct       sfc       sfs       sab
      &X1T      &X1C      &X2T      &X2C      &X3T      &SFC      &SFS      &S12
$#     sbc       sca      sffc    amodel      phic    e_limt    s_delm
      &S23      &S31     &SFFC       2.0     &PHIC   &ELIMIT   &SDELAM
$#   omgmx     ecrsh     eexpn   cr1/sig   am1/fda
    &OMGMX    &ECRSH    &EEXPN   &CR1SIG   &AM1FD1
$# am2/fdb am3/sfc/s am4/mc/de cr2/ea/eb     cr3/g    cr4/ec
   &AM2FD2  &AM3FCPS  &AM4MCDE  &CR2E1E2     &CR3G    &CR4E3
$--------+---------+---------+---------+---------+---------+---------+---------+
$ *** END OF PARAMETER EXPRESSIONS ***
$--------+---------+---------+---------+---------+---------+---------+---------+
*INCLUDE
$--------+---------+---------+---------+---------+---------+---------+---------+
002-contact-SMP.key
$--------+---------+---------+---------+---------+---------+---------+---------+
*INCLUDE
$--------+---------+---------+---------+---------+---------+---------+---------+
$# Use any one of the following FILES with corresponding SPC file
$# The First FILE has a total of 40x40x4=6400 Elements
$# The Second FILE has a total of 80x80x10=64000
$--------+---------+---------+---------+---------+---------+---------+---------+
$# 003-101-102-80mmx80mmx2mm-Composite-Plate-FEM.key
003-101-102-80mmx80mmx2mm-Composite-Plate-FINE-FEM.key
$--------+---------+---------+---------+---------+---------+---------+---------+
*INCLUDE
$--------+---------+---------+---------+---------+---------+---------+---------+
$# 005-1000-SPC.key
005-1000-FINE-SPC.key
$--------+---------+---------+---------+---------+---------+---------+---------+
*INCLUDE
$--------+---------+---------+---------+---------+---------+---------+---------+
004-700-Sphere-FEM.key
$--------+---------+---------+---------+---------+---------+---------+---------+
*PART_MOVE
$-----+-+---------+---+-----+---------+---------+-----+---+---+-----+---------+
$    pid            xmov            ymov            zmov     cid
     700             &dx             &dy             &dz
$--------+---------+---------+---------+---------+---------+---------+---------+
*CONTROL_ENERGY
$--------+---------+---------+---------+---------+---------+---------+---------+
$#    hgen      rwen    slnten     rylen
         2         2         2         2
$--------+---------+---------+---------+---------+---------+---------+---------+
*CONTROL_HOURGLASS
$--------+---------+---------+---------+---------+---------+---------+---------+
$#     ihq        qh
         7  0.100000
$--------+---------+---------+---------+---------+---------+---------+---------+
*CONTROL_OUTPUT
$--------+---------+---------+---------+---------+---------+---------+---------+
$#   npopt    neecho    nrefup    iaccop     opifs    ipnint    ikedit    iflush
         0         0         0         0     0.000         0       100       100
$#   iprtf    ierode     tet10    msgmax    ipcurv
```

```
         0         1         2        50
$--------+---------+---------+---------+---------+---------+---------+---------+
*CONTROL_TERMINATION
$--------+---------+---------+---------+---------+---------+---------+---------+
$#  endtim    endcyc     dtmin    endeng    endmas
  &ENDTIME
$--------+---------+---------+---------+---------+---------+---------+---------+
*CONTROL_TIMESTEP
$--------+---------+---------+---------+---------+---------+---------+---------+
$#  dtinit    tssfac      isdo    tslimt     dt2ms      lctm     erode     ms1st
     0.000   &TSSFAC
$#  dt2msf   dt2mslc     imscl
     0.000         0         0
$--------+---------+---------+---------+---------+---------+---------+---------+
*DATABASE_GLSTAT
$--------+---------+---------+---------+---------+---------+---------+---------+
$#      dt    binary      lcur     ioopt     dthff     binhf
    &DTBIN         3
$--------+---------+---------+---------+---------+---------+---------+---------+
*DATABASE_MATSUM
$--------+---------+---------+---------+---------+---------+---------+---------+
$#      dt    binary      lcur     ioopt     dthff     binhf
    &DTBIN         3
$--------+---------+---------+---------+---------+---------+---------+---------+
*DATABASE_RCFORC
$--------+---------+---------+---------+---------+---------+---------+---------+
$#      dt    binary      lcur     ioopt     dthff     binhf
    &DTBIN         3
$--------+---------+---------+---------+---------+---------+---------+---------+
*DATABASE_BINARY_D3PLOT
$--------+---------+---------+---------+---------+---------+---------+---------+
$#      dt      lcdt      beam     npltc
     &DTD3         0
$#   ioopt
         0
$--------+---------+---------+---------+---------+---------+---------+---------+
*DATABASE_EXTENT_BINARY
$--------+---------+---------+---------+---------+---------+---------+---------+
$#   neiph     neips    maxint    strflg    sigflg    epsflg    rltflg    engflg
    &NEIPH    &NEIPH         3         1         1         1         1         1
$#  cmpflg    ieverp    beamip     dcomp      shge     stssz    n3thdt   ialemat
         0         1         0         1         1         1         2         1
$# nintsld   pkp_sen      sclp    unused     msscl     therm
         0         0  1.000000
$--------+---------+---------+---------+---------+---------+---------+---------+
*CONTACT_FORCE_TRANSDUCER_PENALTY_ID
$--------+---------+---------+---------+---------+---------+---------+---------+
$#     cid title
       700 Sphere Impact Force
$--------+---------+---------+---------+---------+---------+---------+---------+
$#    ssid      msid     sstyp     mstyp    sboxid    mboxid       spr       mpr
       700                   3
$#      fs        fd        dc        vc       vdc    penchk        bt        dt
     0.000     0.000     0.000     0.000     0.000         0     0.000  1.00E+20
$#     sfs       sfm       sst       mst      sfst      sfmt       fsf       vsf
  0.000000  0.000000     0.000     0.000  0.000000  0.000000  0.000000  0.000000
$--------+---------+---------+---------+---------+---------+---------+---------+
*SET_PART_LIST
$--------+---------+---------+---------+---------+---------+---------+---------+
$#     sid       da1       da2       da3       da4
       123
$#    pid1      pid2      pid3      pid4      pid5      pid6      pid7      pid8
       101       102       700
$--------+---------+---------+---------+---------+---------+---------+---------+
*SET_PART_LIST
$--------+---------+---------+---------+---------+---------+---------+---------+
$#     sid       da1       da2       da3       da4
       700
$#    pid1      pid2      pid3      pid4      pid5      pid6      pid7      pid8
       700
```

```
$--------+---------+---------+---------+---------+---------+---------+---------+
*PART
$--------+---------+---------+---------+---------+---------+---------+---------+
Composite Layer 1 TOP
$#     pid     secid       mid     eosid      hgid      grav    adpopt      tmid
       101       100       101
$--------+---------+---------+---------+---------+---------+---------+---------+
*PART
$--------+---------+---------+---------+---------+---------+---------+---------+
Composite Layer 2 BOTTOM
$#     pid     secid       mid     eosid      hgid      grav    adpopt      tmid
       102       100       102
$--------+---------+---------+---------+---------+---------+---------+---------+
*PART
$--------+---------+---------+---------+---------+---------+---------+---------+
Sphere Projectile
$#     pid     secid       mid     eosid      hgid      grav    adpopt      tmid
       700       100       700
$--------+---------+---------+---------+---------+---------+---------+---------+
*SECTION_SOLID
$--------+---------+---------+---------+---------+---------+---------+---------+
$#   secid    elform       aet
       100         1
$--------+---------+---------+---------+---------+---------+---------+---------+
*INITIAL_VELOCITY_GENERATION
$--------+---------+---------+---------+---------+---------+---------+---------+
$#      id      styp     omega        vx        vy        vz     ivtan
       700         2       0.0       &VX       &VY       &VZ         0
$       xc        yc        zc        nx        ny        nz     phase

$--------+---------+---------+---------+---------+---------+---------+---------+
*END
$--------+---------+---------+---------+---------+---------+---------+---------+
```

FILE: 002-contact-SMP.key

```
$--------+---------+---------+---------+---------+---------+---------+---------+
*KEYWORD
$--------+---------+---------+---------+---------+---------+---------+---------+
*CONTROL_CONTACT
$--------+---------+---------+---------+---------+---------+---------+---------+
$#  slsfac    rwpnal    islchk    shlthk    penopt    thkchg     orien    enmass
   &SLSFAC     0.000         1         0         0         0         1         1
$#  usrstr    usrfrc     nsbcs    interm     xpene     ssthk      ecdt   tiedprj
         0         0    &BSORT         0  4.000000
$#   sfric     dfric       edc       vfc        th     th_sf    pen_sf
     0.000     0.000     0.000     0.000     0.000     0.000     0.000
$#  ignore    frceng   skiprwg    outseg   spotstp   spotdel   spothin
         1         0         0         0         0         0     0.000
$#    isym    nserod    rwgaps    rwgdth     rwksf      icov         x    ithoff
         1         0         0     0.000  1.000000         0  1.000000
$--------+---------+---------+---------+---------+---------+---------+---------+
*CONTACT_ERODING_SINGLE_SURFACE
$--------+---------+---------+---------+---------+---------+---------+---------+
$#    ssid      msid     sstyp     mstyp    sboxid    mboxid       spr       mpr
       123                   2
$#      fs        fd        dc        vc       vdc    penchk        bt        dt
     0.500     0.300     0.000     0.000     20.00         0      0.00  1.00E+28
$#     sfs       sfm       sst       mst      sfst      sfmt       fsf       vsf
  1.000000  1.000000     0.000     0.000  0.000000  0.000000  0.000000  0.000000
$#    isym    erosop      iadj
         0         1         1
$#    soft    sofscl    lcidab    maxpar     sbopt     depth     bsort    frcfrq
         2   &SLSFAC         0   &PARMAX    &SBOPT    &DEPTH    &BSORT         1
$#  penmax    thkopt    shlthk     snlog      isym     i2d3d    sldthk    sldstf
                                                 0         1   &SLDTHK     0.000
$#    igap    ignore    dprfac    dtstif    unused    unused    flangl
         1         1
$--------+---------+---------+---------+---------+---------+---------+---------+
*END
$--------+---------+---------+---------+---------+---------+---------+---------+
```

FILE: 003-101-102-80mmx80mmx2mm-Composite-Plate-FEM.key

This file contains the *NODE & *ELEMENT_SOLID cards for the composite parts 101 and 102.

FILE: 004-700-Sphere-FEM.key

This file contains the *NODE & *ELEMENT_SOLID cards for the Steel sphere part 700.

FILE: 005-1000-SPC.key

```
$--------+---------+---------+---------+---------+---------+---------+---------+
*KEYWORD
$--------+---------+---------+---------+---------+---------+---------+---------+
*BOUNDARY_SPC_SET
$#    nsid       cid      dofx      dofy      dofz     dofrx     dofry     dofrz
      1000         0         1         1         1         1         1         1
*SET_NODE_LIST_TITLE
Edge Fixed SPC
$#     sid       da1       da2       da3       da4    solver
      1000     0.000     0.000     0.000     0.000MECH
$#    nid1      nid2      nid3      nid4      nid5      nid6      nid7      nid8
   1000001   1000002   1000003   1000004   1000005   1000006   1000007   1000008
   1000009   1000010   1000011   1000012   1000013   1000014   1000015   1000016
.
.
.
   1008390   1008391   1008392   1008393   1008394   1008395   1008396   1008397
   1008398   1008399   1008400   1008401   1008402   1008403   1008404   1008405
$--------+---------+---------+---------+---------+---------+---------+---------+
*END
$--------+---------+---------+---------+---------+---------+---------+---------+
```

RUNNING A LARGER MODEL

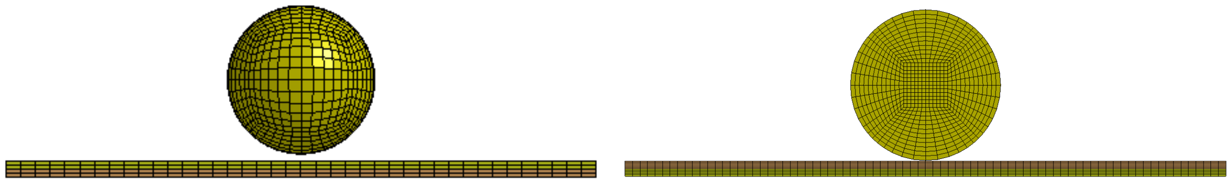
Above files runs a smaller model with 40x40x4=6400 elements for the composite plate. There are two additional files that will run a larger model with 80x80x10=64000 elements for the composite plate. Name of these two files are as follows:

003-101-102-80mmx80mmx2mm-Composite-Plate-FINE-FEM.key
005-1000-FINE-SPC.key

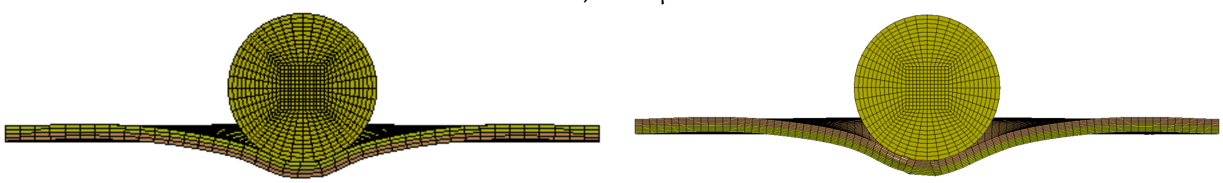
RESULTS AND DISCUSSION

Deformation as a Function of Time

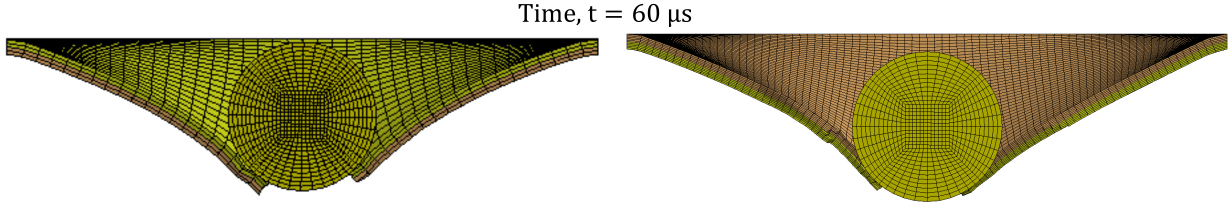
SMALL MODEL with 6400 Elements **LARGE MODEL with 64000 Elements**



Time, t = 0 μs



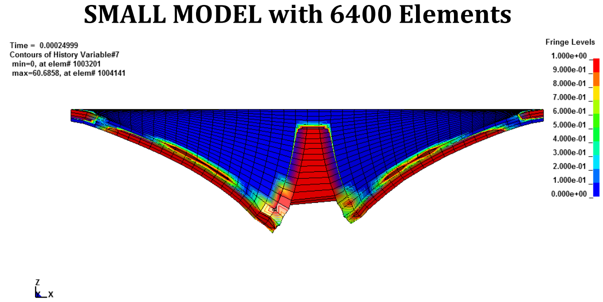
Time, t = 60 μs



Time, t = 250 μs Time, t = 260 μs

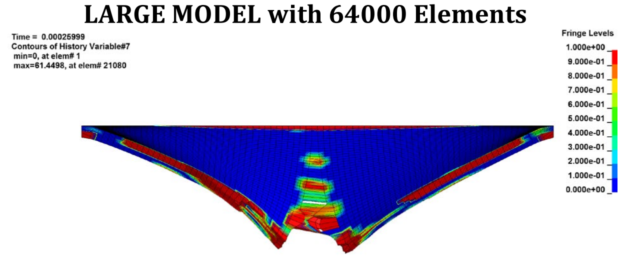
Contour of History Variable # 7, Fiber Tension-Shear Damage Mode along A

SMALL MODEL with 6400 Elements



Time, t = 250 μs

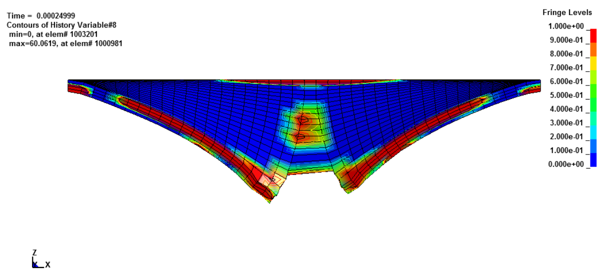
LARGE MODEL with 64000 Elements



Time, t = 260 μs

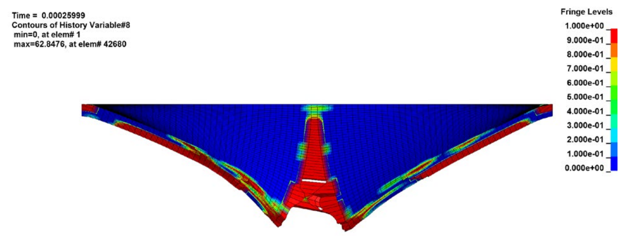
Contour of History Variable # 8, Fiber Tension-Shear Damage Mode along B

SMALL MODEL with 6400 Elements



Time, t = 250 μs

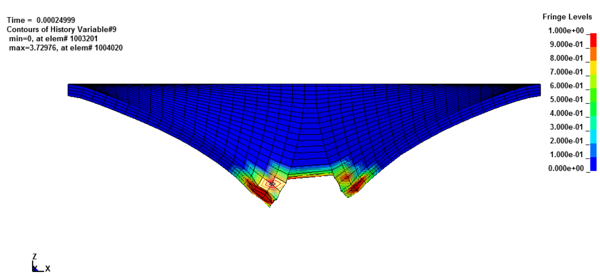
LARGE MODEL with 64000 Elements



Time, t = 260 μs

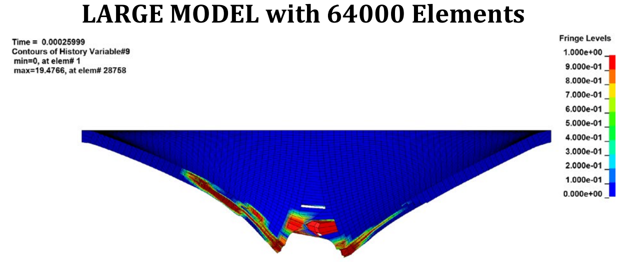
Contour of History Variable # 9, Fiber Crush Damage Mode

SMALL MODEL with 6400 Elements



Time, t = 250 μs

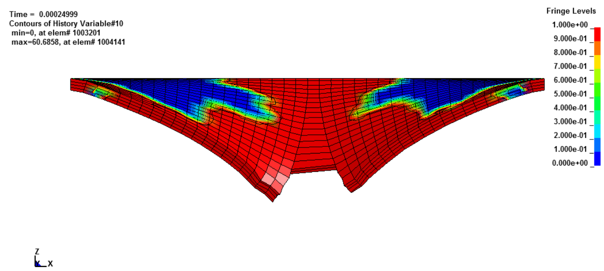
LARGE MODEL with 64000 Elements



Time, t = 260 μs

Contour of History Variable # 10, In-Plane Matrix Damage Mode

SMALL MODEL with 6400 Elements



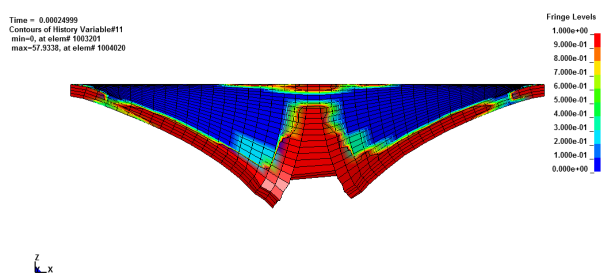
Time, t = 250 μs

LARGE MODEL with 64000 Elements

Time, t = 260 μs

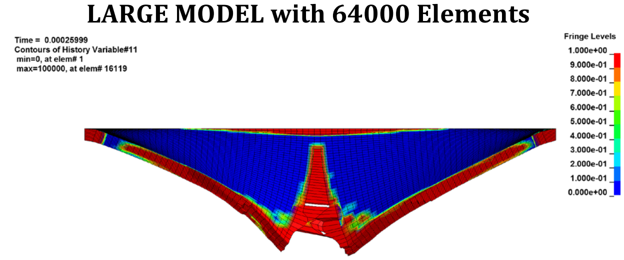
Contour of History Variable # 11, Transverse Matrix Damage Mode

SMALL MODEL with 6400 Elements



Time, t = 250 μs

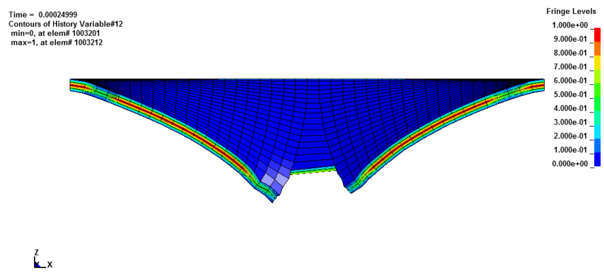
LARGE MODEL with 64000 Elements



Time, t = 260 μs

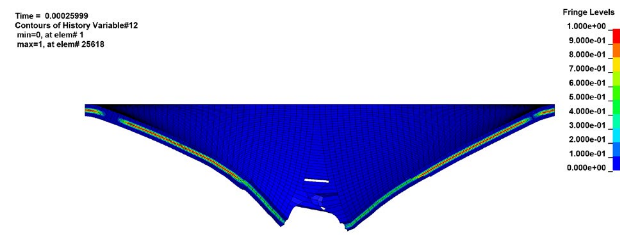
Contour of History Variable # 12, Delamination Damage Mode

SMALL MODEL with 6400 Elements



Time, t = 250 μs

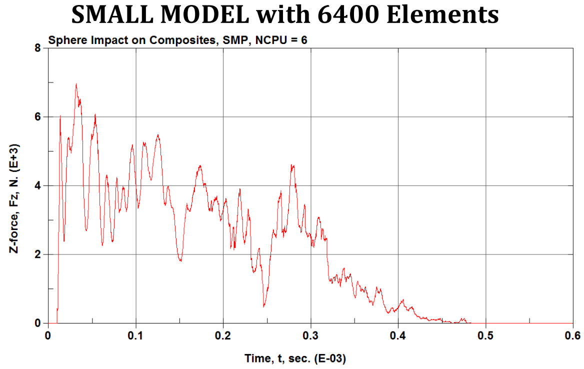
LARGE MODEL with 64000 Elements



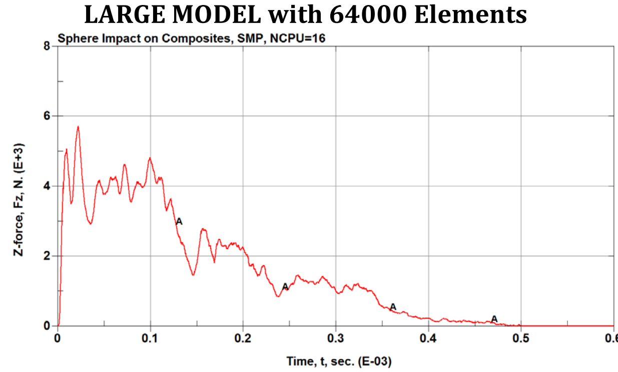
Time, t = 260 μs

Time History of Impact-Contact Force

SMALL MODEL with 6400 Elements



LARGE MODEL with 64000 Elements



This example problem can be downloaded from UD-CCM MAT162 website:
http://www.ccm.udel.edu/software/mat162/

EXAMPLE 1: Sphere-Impact-on-Composites-SOLID-MAT162.zip

```
This zip folder contains the following KEWORD files.  Run the first file.
001-Sphere-Impact-on-Composite-Plate.key
002-contact-SMP.key
003-101-102-80mmx80mmx2mm-Composite-Plate-FEM.key
003-101-102-80mmx80mmx2mm-Composite-Plate-FINE-FEM.key
004-700-Sphere-FEM.key
005-1000-FINE-SPC.key
005-1000-SPC.key
```

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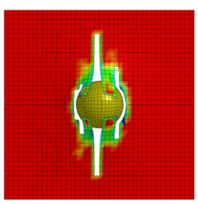
EXAMPLE 2: SPHERE IMPACT ON COMPOSITE SHELLS & SOLIDS
Comparing Composite Damage Modeling with MAT-162 and MAT-054

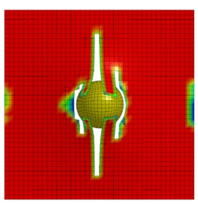
MAT162 is a rate dependent progressive composite damage model for solid elements. In composite shell elements, transverse shear damage cannot be defined and this is why a composite solid element defining transverse shear damage is essential in modeling problems where transverse shear damage is important. Shell elements are efficient for thin-section composites, whereas thick-section composites can be modeled using multiple through-thickness elements. MAT162 for solid elements have unique quadratic progressive damage modes, which no other material model offers. These unique damage modes, such as, (i) Tension-Shear, (ii) Axial Compression, (iii) Inter laminar Shear (ILS) and Delamination, (iv) Fiber Shear (FS) or Punch Shear (PS), and (v) Fiber Crush provides the theoretical framework for modeling most known composite damages.

Benefits of MAT162 progressive damage can be elucidated with simple examples. Consider the transverse impact of a 20mm sphere with an impact velocity of 50 m/s on an 80mm x 80mm x 2mm composite plate modeled with perfectly clamped boundary condition. Shell elements with MAT054 and solid elements with MAT162 damage predictions shows that MAT054 can only model in-plane tension and compression damage, while MAT162 can predict tension-shear along direction 1 & 2, in-plane shear, transverse shear and associated delamination.

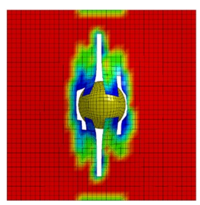
Damage Modes: Shell Elements – MAT054



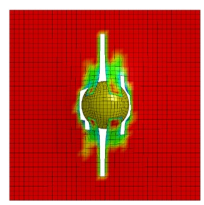
HISV 1:
Tensile Fiber Mode



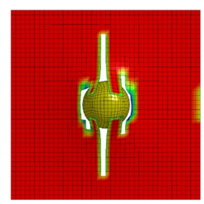
HISV 2:
Compres. Fiber Mode



HISV 3:
Tensile Matrix Mode

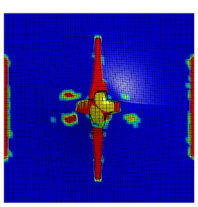


HISV 4:
Compres. Matrix Mode

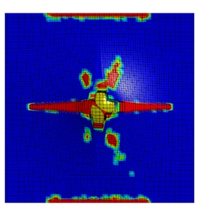


HISV 5:
max[ef(ip)]

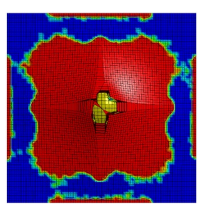
Damage Modes: Solid Elements – MAT162



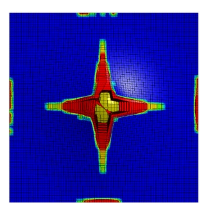
HISV 7:
Tension-Shear Mode 1



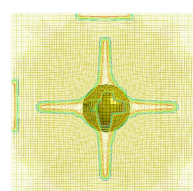
HISV 8:
Tension-Shear Mode 2



HISV 10:
In-Plane Matrix Mode



HISV 11:
Transverse Matrix Mode



HISV 12:
Delamination Index

In modeling out-of-plane deformation of any kind, the primary damage mode is tension-shear where MAT162 defines a quadratic damage function with in-plane tension and out-of-plane shear properties for the initiation of tension-shear damage.

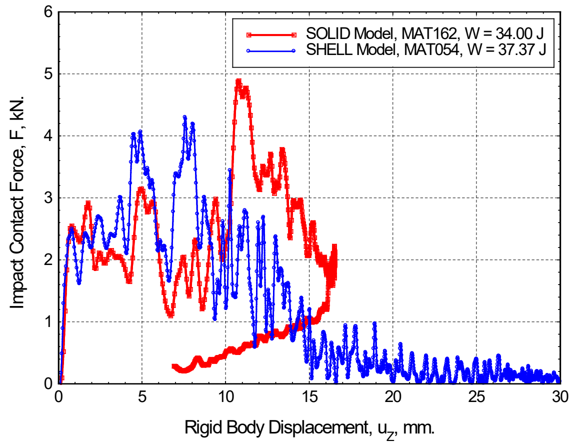
$$f_7 = \left\{\frac{E_1\langle\varepsilon_1\rangle}{X_1^T}\right\}^2 + \left\{\frac{G_{31}\varepsilon_{31}}{S_1^{FS}}\right\}^2$$

The MAT054 equivalent damage function uses in-plane tension and in-plane shear properties instead, and in general fails to account for transverse shear deformation.

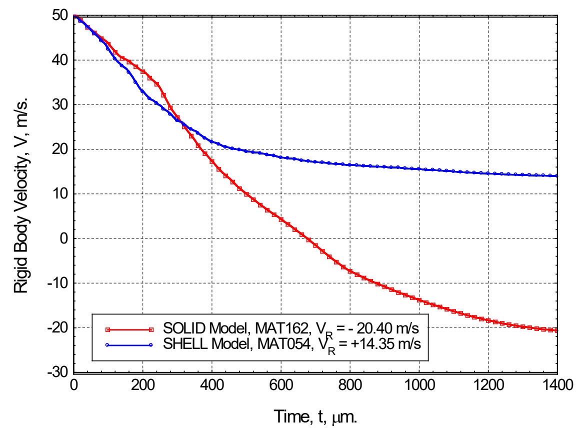
$$e_f = \left\{\frac{E_1\varepsilon_1}{X_1^T}\right\}^2 + \beta\left\{\frac{G_{12}\varepsilon_{12}}{S_{12}}\right\}$$

The inability of MAT54 in modeling transverse damage modes greatly affects the impact-contact force and associated work done, predicts premature perforation while MAT162 predicts partial perforation and complete rebound of the sphere in case of initial impact velocity of 50 m/s.

Quantitative Comparison between MAT054 (Shell) & MAT162 (Solid) Material Models



Force-Displacement



Time-History of Rigid Body Velocity

A local MAT162 solid model in the anticipated damage region combined with global shell elements can be used to efficiently and accurately model large-scale impact application such as, crash analysis of automotive composite components and aerospace structures under impact conditions.

This example problem can be downloaded from UD-CCM MAT162 website:
http://www.ccm.udel.edu/software/mat162/

EXAMPLE 1: Sphere-Impact-on-Composites-SOLID-MAT162.zip

```
This zip folder contains the following KEWORD files.  Run the first file.
001-Sphere-Impact-on-Composite-Plate.key
002-contact-SMP.key
003-101-102-80mmx80mmx2mm-Composite-Plate-FEM.key
003-101-102-80mmx80mmx2mm-Composite-Plate-FINE-FEM.key
004-700-Sphere-FEM.key
005-1000-FINE-SPC.key
005-1000-SPC.key
```

EXAMPLE 2: Sphere-Impact-on-Composites-SHELL-MAT54.zip

```
This zip folder contains the following KEWORD files.  Run the first file.
001-Sphere-Impact-on-Composite-Shell.key
002-control-cards.key
003-Shell-Plate-80mm-x-80-mm-FEM.key
004-700-Sphere-FEM.key
005-parts-partsets-sections-materials-FEM.key
006-100-Node-SET.key
006-nodeset-elementset-segmentset-FEM.key
007-contact-single-surface.key
008-boundary-conditions.key
009-initial-conditions.key
010-loading-conditions.key
011-output-database.key
```