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# **NIVERSITY**OF **ELAWARE**

# CDS: Composite Design Software Next Generation Interface for Design and Analysis of Composite Structures

# **MATERIAL INPUT**

terials La	nina	Laminat	e Struct	ure Loadin	g Sour	ces	Scaling
New Mate	rial :			Export to:	Abaqus	Ansys WB	SimDesigner
View	Upda	te 🔲 Au	to Update Delet	Clear	Lamina 💌	Open	Save
Mechanical	Physical	Mi	icromechanics	Cure Kinetics	Failur	e	Non-Linear
Reduced Properties	Max Stree	ss Ma	x Strain Hydr	ostatic Tsai-Wu	Christenser	Feng	Hashin
			Enter Properties	Sources		🔺 Rec	luced Property
Longitudinal Young's Modul	us (Tension): E1T						E 🔺 🛛
Longitudinal Young's Modul	us (Compression)	): E1T					J 🛒 76
Transverse Young's Modulu	s (Tension): E2T						Apply
Transverse Young's Modulu	s (Compression)	: E2T					Chdd.
Transverse Young's Modulu	s (Tension): E3T						
- Transverse Young's Modulu	s (Compression)	: E3T					
Poissons Ratio: v12							
Poissons Ratio: v13							
Poisson's Ratio: v23							
Shear Modulus: G12							
Charas Madulues, C42							

Atterial input for CDS includes the following properties

- Mechanical Properties, micromechanics input
- Physical Properties, cost
- Failure properties, reduced property sets
- Non-linear properties, MAT162 Property lists

### **EFFECTIVE PROPERTIES**

w Current 🛛 🖌 Lamina	te 🗸 Select [0]	0	Add to	: My Mal	terials 💌 Na	ame		Save
operties	Resultants		St	resses/St	rains	ſ	Failure	
urrent Thin Section Prope	rties	וו			O	oserve Ply 🕂	]	1
	Result	AB	D Matrix					
Tensile Modulus X-Direction: Ex	4172329.250000		748975	246918	l 0	0	0	0
Tensile Modulus Y-Direction: Ey	4172329.250000		246918	748975	0	0	0	0
Poisson's Ratio: vxy	0.329674		0	0	251029	0	0	0
Poisson's Ratio: vyx	0.329674		0	0	0	1802	446	21
Shear Modulus: Gxy	1568929.125000		0	0	0	446	1556	21
hermal Expansion Coeff: ax	0.000000		0	0	0	21	21	454
hermal Expansion Coeff: ay	0.000000		0	v		21	21	404
Thermal Expansion Coeff: axy	0.000000	ab	cd Matrix					
Moisture Expansion Coeff: bx	0.000000		1.50E-6	-493.84E-9	-146.49E-15	0.00E+0	0.00E+0	0.00E+0
Moisture Expansion Coeff: by	0.000000	-4	493.84E-9	1.50E-6	-2.57E-15	0.00E+0	0.00E+0	0.00E+0
Flexural Stiffness: Dx	1674.376099	-1	46.49E-15	-2.57E-15	3.98E-6	0.00E+0	0.00E+0	0.00E+0
ending Stiffness: E <sup>x</sup> I = 1 *Dx	3348.752197		0.00E+0	0.00E+0	0.00E+0	597.24E-6	-170.80E-6	-19.31E-6
Node 1 Frequency: w1	16.267310		0.00E+0	0.00E+0	0.00E+0	-170.80E-6	692.08E-6	-23.61E-6
vlode 1 Frequency: w2	65.069239		0.00E+0	0.00E+0	0.00E+0	-19.31E-6	-23.61E-6	2.20E-3
vlode 1 Frequency: w3	146.405787	0	Matrix for	nlv	•	0 Prime M	atrix for this	nlv
			1093264	274306	0 1	11093264	274306	1 0
			274306	806783	0	274306	806783	0
			0	0	300000	0	0	300000

Outputs results for:

- Thin or thick section effective properties
- Load and Strain Resultants



Outputs include internal stresses, strains, displacements and factors of safety from mechanical, thermal or moisture loading for thin, thick walled plates or cylinders

### (Continued)

terials	Lamina	Lamin	ate	Stru	ucti	ure	Load	Cases	Sou	irces	S	caling		
New Lai	minate: [0/90/45	/-45]2s		Lar	mina	ate Ty	pe Thin	Laminat	e 🗸	Open		Save		
View	Add	date 🗌	Auto L	Jpdate	De	elete	Clear		Fill	Symmet	ric	Undo	51	
						_				-				
rkbench	Hide Workbeng	h Number	of Plies	16		La	minate	View S	ummary		Angle		~	
	Value		2222				Mater	ial	Thickness	Angle	рт 🛛	DM	~	
Resulting Thickness 0.800000 =n <sup>-</sup> t stial Bly Thickness 0.050			1	Kevlar L	amina	0.0100	0	0	0					
			2	Kevlar Lamina	amina	0.0100	90 0	0	0					
al Angle	0		IICKI IESS		=1	3	Kevlar L	amina	0.0100	45	45 0	0		
al Delta Temperatu	Delta Temperature 0			4	Kevlar Lamina	amina	0.0100	-45	0	0				
al Delta Moisture	0	Pop	ulate>	< Grah		5	Kevlar L	amina	0.0100	-45	0	0		
						6	Kevlar L	amina	0.0100	45	0	0		
Fill Symn	netric Clear	4 PI	ies 🗸	Laminate	e >	7	Kevlar L	amina	0.0100	90	0	0		
Material	Thickness	Angle I	DT			8	Kevlar L	amina	0.0100	0	0	0		
					÷.	9	Kevlar L	amina	0.0100	0	0	0		
II Insert Material						10	Kevlar L	amina	0.0100	90	0	0		
Insert Material						11	Kevlar L	amina	0.0100	45	0	0		
Insert Material		I												

Laminate input for CDS includes the following:

- Material Selection, thickness, angle, ply delta temperature and moisture, and winding tension (for cylinders)
- Workbench allows for rapid creation of multiple laminates for design studies



# **PROGRESSIVE FAILURE**

esults vi	ew Current	Thin	- Structu	re 💌 Sele	et			1
	New	Mat.			Add to:	My Ma	terials 💌	Save
roperties		Resultants		Stresses/S	Strains		Failure	
ailure Criteria	Max Stre	55 • Direction						_
View	Load v's Strain	💌 <sup>r</sup> -Direction		Material	Failure Mode	A-m-mane X-Load	X-Direction	
Plo	t In Plane	e W-Directi		Kevlar Lamina	XIT.	2500.	0.0036	-1
5000-			2	Keyler Lemine				_
3000			3	KevlarLamina	XOTT,	2600.	0.0037,	_
4000			4	KevlarLamina	XY1T.	2600.	0.0037.	
4000-			S	Keylar Lamina	XYIT,	2600,	0.0037,	_
			6	KevlarLamina	XOLT.	2600.	0.0037.	
8 3000-			7	Keyler Lemine				
8			8	KevlarLamina	XIT.	2900.	0.0036	
S 2000-			9	Keyler Lemine	X17,	2500,	0.0036	
	1		10	KevlarLamina				
1000 -			11	KevlarLamina	XYIT.	2600.	0.0037.	
	1		12	Kevlar Lamina	XYLT,	2600,	0.0037,	
0-			13	KevlarLamina	XOTET.	2600.	0.0037.	
-0.02	5 0	0.025 0.05	0.075 14	Keylar Lamina	XYLT.	2600.	0.0037.	
Linear w		Strain	15	KevlarLamina				
			1.6	Manuface I recently as	1911	25.00	0.0024	_

- Progressive failure using max, stress and strain, Tsai-Wu failure results under multiaxial loading
- Outputs include, failure ply and mode, loadstrain plots, property reduction over loading



## **STRUCTURES**



in Section Composites



**Thick Section Composites** 



Thick Walled Cylinders



**Hybrid Structures** 

### **CDS AVAILABILITY**

Software available to current Industrial Consortium members, university researchers and collaborating government agencies

### ACKNOWLEDGEMENTS

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